

PREPARED	PIPER AIRCRAFT CORP. DEVELOPMENT CENTER, VERO BEACH, FLA.	VB163
CHECKED		
APPROVED		PAGE <u>I</u>

DUPLICATE

AIRPLANE FLIGHT MANUAL

MODEL PA-28-180

Serial numbers 761 Thru 5600

FAA IDENTIFICATION NO. EC-IDP  
SE-EPY

SERIAL NO. 28-2768

C-FZPX S/N 28-1305

THIS DOCUMENT MUST BE KEPT IN AIRPLANE, AT ALL TIMES.

FAA APPROVED: Original signed by Walter R. Haldeman \*  
Walter R. Haldeman  
Chief, Engineering & Manufacturing Branch  
Southern Region - - - Atlanta, Georgia

DATE: August 3, 1962

FAA APPROVED: Gene Dearing For Retype Only.  
Gene Dearing  
Aerospace Engineer

DATE: August 12, 1964

Ent. LVD 2442 Mont 1/4-93/  
Soren Hansson  
NH-470807-P235

Log of Revisions

<u>REVISION NO.</u>	<u>PAGE</u>	<u>DESCRIPTION</u>	<u>APPROVED</u>	<u>DATE</u>
1	1	Deleted Propeller Pitch Information. Added Static R.P.M. Information	<i>J. A. Ragan</i> for H. E. Waterman Supervisor SO-EMDO-42	5/25/64
2	2	Placards Section: Added Placard No. 5	<i>H. E. Waterman</i> Supervisor SO-EMDO-42	7/8/64
3	2	Added to Placard No. 3: "BAGGAGE, MAX. 200 LBS., SEE WEIGHT AND BALANCE DATA FOR BAGGAGE LOADINGS BETWEEN 150 LBS. AND 200 LBS."	<i>Robert S. Sauer</i> for H. C. Faller Supervisor SO-EMDO-43	8/5/64
	1	Added Sensenich M76EMMS		
4	3	Item 5 added to Procedures Section.	<i>H. C. Faller</i> Supervisor SO-EMDO-43	10/20/64
5	1	Limitations Section: Revised Oil Temperature and Fuel Pressure Range	<i>Robert S. Sauer</i> for H. C. Faller Supervisor, SO-EMDO-43	6/23/65
6	1	Limitation Section: Add note to Engine Limits	<i>H. C. Faller</i> Supervisor, SO-EMDO-43	1/5/66
7	2	C. G. Range: 1975 lbs. 85.9 in. 95.9 in. 1650 lbs. 84.0 in. 95.9 in. Was 18,50 lbs. 85.1 in. 95.9 in.		
	4	Added Procedures Section And Item 6		
	2	Added Placard No. 6	<i>Robert S. Sauer</i> for H. C. Faller Supervisor SO-EMDO-43	5/20/66

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DEVELOPMENT CENTER, VERO BEACH, FLA.

Airplane Flight Manual  
Model PA-28-180

REPORT VE-163

PAGE III

Log of Revisions

<u>Revision No.</u>	<u>Page</u>	<u>Description</u>	<u>Approved</u>	<u>Date</u>
8	1	Revised Oil Temperature, Oil Pressure and Fuel Pressure Limitations		
	2,3	Revised Placards No. 3 and No. 5		
	5	Added Page 5		
		Procedures Section - Added Item 7		
	6	Added Page 6	<i>Henry C. Faller</i> Henry C. Faller Supervisor SO-EMDO-43	7/15/66
9	1	Limitations Section Add "or O-360-A4A	<i>Henry C. Faller</i> Henry C. Faller Supervisor SO-EMDO-43	8/2/66
10	2,3	C. G. Range - Placard No. 1 and Placard No. 3 revised to include utility category operations. Added utility category max. wt. and approved maneuvers		
	4	Procedures Section - Added to Item 3 "For Normal Category Operation": Added Placard No. 7.		
	3	Placards Section - Added utility category operation to Item 4.		
	1	Added Utility Category		
	2	Added maximum positive load factor for Utility Category. Added Baggage Capacity.	<i>Henry C. Faller</i> Henry C. Faller Supervisor SO-EMDO-43	12/16/66

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REVISION NO.	PAGE	DESCRIPTION	APPROVED	DATE
11	3	Placards Section: Revised Placard No. 1 to read, "In Full View of the Pilot"	<i>H.C. Faller</i> H. C. Faller Supervisor SO-EMDO-43	5/12/67
12	2	Revised C. G. Range	<i>H.C. Faller</i> H. C. Faller Supervisor SO-EMDO-43	2/25/67
13	3, 4	Revised Placard No. 4 and No. 7 to read: "In full view of the pilot"	<i>H.C. Faller</i> H. C. Faller Supervisor SO-EMDO-43	4/2/68
14	1	Added Aircraft Serial Numbers 1571 and 1573 to Engine and Propeller Limitations	<i>H.C. Faller</i> H. C. Faller Supervisor SO-EMDO-43	6/3/68
15	1	Added Propeller Designations	<i>H.C. Faller</i> H. C. Faller Supervisor SO-EMDO-43	6/24/68
16	Title	Allocated Piper Report No. VB-163 to this Manual.	<i>H.M. Toomey</i> Herb M. Toomey FAA DOA SO-1	11/14/68
17	Title	Added Applicable Serial Nos. 1 Thru 4377	<i>H.M. Toomey</i> H. M. Toomey FAA DOA SO-1	4/22/69
	1	Added Supplement No. 1		

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Log of Revisions

REVISION NO.	PAGE	DESCRIPTION	APPROVED	DATE
18	Title	Changed applicable Serial Nos. from 1 thru 4377 to 1 thru 5600.	<i>H. M. Toomey</i> H. M. Toomey FAA DOA SO-1	7/15/69
19	Title	Changed applicable Serial Nos. from 1 thru 5500 to 1 thru 5600.	<i>H. M. Toomey</i> H. M. Toomey FAA DOA SO-1	9/23/69
20	2	Added Forward Intermediate and Forward Gross Weight Points	<i>H. M. Toomey</i> H. M. Toomey FAA DOA SO-1	5/8/70
21	2	Deleted Forward Intermediate and Forward Gross Weight Points	<i>G. C. Stephen</i> G. C. Stephen FAA DOA SO-1	9/14/70
22	1	Changed oil pressure gauge markings	<i>Ward Evans</i>	7-25-75

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SUPPLEMENT NO. 1 TO PIPER MODEL PA-28 FLIGHT MANUAL

MODELS AFFECTED: Piper PA-28 models equipped with  
Lycoming O-360-A3A engine and  
Sensenich M76EMM-0, M76EMMS-0,  
76EM8S5-0 or 76EM8-0 propeller.

PROPELLER LIMITS

Avoid continuous operation between 2150 and 2350 RPM.

The aircraft tachometer must be placarded to show a red arc between 2150 and 2350 RPM in accordance with Piper Service Letter No. 526.

NOTE: This document must be attached to the Airplane Flight Manual.

FAA DOA SO-1  
APPROVED

H. M. Toomey  
H. M. Toomey

DATE

4/22/69



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Piper Model PA-28-180.  
Normal and Utility Categories

AIRPLANE FLIGHT MANUAL

1. Limitations Section      The following limitations must be observed in the operation of this airplane.
- Engine      Lycoming O-360-A3A or O-360-A4A      **FZPX - O-360-A4A**
- Engine Limits      Maximum permissible RPM for takeoff, 2475. For all other operations, 2700 rpm, 180 hp, (A/C S/N 28-671 to 1760A).  
For all operations, 2700 rpm, 180 hp, (A/C S/N 28-1571, 1573, 1761 and up).
- Fuel      91/96 minimum octane aviation fuel.  
**FZPX - 76EM8-0-60**
- Propeller      Sensenich M76 EMM or 76EM8 (S/N 671 to 1760A)  
Sensenich M76 EMMS or 76EM8S5 (S/N 1571, 1573, 1761 & up).  
Maximum diameter 76 inches, minimum diameter 76 inches.  
Static RPM at maximum permissible throttle setting. Not over 2450, not under 2275. No additional tolerance permitted.
- Power Instruments      Oil temperature: GREEN arc (normal operating range) 120°F to 245°F; YELLOW arc (caution range) 60°F to 120°F; RED line (maximum) 245°F (S/N 671 to S/N 1760A)
- Oil Temperature: GREEN arc (normal operating range) 75°F to 245°F; RED line (maximum) 245°F (S/N 1571, 1573, 1761 & up).
- Oil Pressure: GREEN arc (normal operating range) 60 psi to 90 psi; YELLOW ARC (caution range) 25 psi to 60 psi; RED line (minimum) 25 psi when installed or 60 psi when installed; RED line (maximum) 90 psi.
- Fuel Pressure: GREEN arc (normal operating range) .5 psi to 5 psi; RED line (minimum) .5 psi; RED line (maximum) 5 psi (S/N 671 to S/N 1760A)
- Fuel Pressure: GREEN arc (normal operating range) .5 psi to 8 psi; RED line (minimum) .5 psi; RED line (maximum) 8 psi (S/N 1571, 1573, 1761 and up)
- Tachometer: GREEN arc (normal operating range) 500 to 2700 rpm; RED line (maximum continuous power) 2700 rpm.

Airspeed Limits	Never exceed ..... 171 mph <span style="color: blue;">148 KT</span> Maximum structural cruise ..... 140 <span style="color: blue;">121</span> Maneuvering ..... 129 <span style="color: blue;">112</span> Flaps extended ..... 115 <span style="color: blue;">100</span> Maximum positive load factor ..... 3.8 Normal Category Maximum positive load factor ..... 4.4 Utility Category Maximum negative load factor ..... No inverted maneuvers approved.	
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Maximum Weight      2400 lbs - Normal Category; 150 lbs - Utility Category.

Baggage Capacity      200 lbs

C. G. Range            The datum used is 78.4 inches ahead of wing leading edge at the intersection of the straight and tapered section.

1. Normal Category

Weight (Pounds)	Forward Limit (In. Aft of Datum)	Rearward Limit (In. Aft of Datum)
2400	92.1	94.5
2200	89.2	95.9
1975	85.9	95.9
1650	84.0	95.9

2. Utility Category

Weight (Pounds)	Forward Limit (In. Aft of Datum)	Rearward Limit (In. Aft of Datum)
1950	85.8	86.5
1650	84.0	86.5

Straight line variation between points given.

NOTE: It is the responsibility of the airplane owner and the pilot to insure that the airplane is properly loaded. See weight and section for proper loading instructions.

- Maneuvers
1. Normal Category - All acrobatic maneuvers including spins prohibited.
  2. Utility Category - Approved maneuvers for Utility Category only.

	<u>Entry Speed</u>
Spins (Flaps Up) .....	Stall
Steep Turns .....	129 mph <span style="color: blue;">112 KT</span>
Lazy Eights .....	129 <span style="color: blue;">112</span>
Chandelles .....	129 <span style="color: blue;">112</span>

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Placards

1. In full view of the pilot:

"THIS AIRPLANE MUST BE OPERATED AS A NORMAL OR UTILITY CATEGORY AIRPLANE IN COMPLIANCE WITH THE OPERATING LIMITATIONS STATED IN THE FORM OF PLACARDS, MARKINGS AND MANUALS.

ALL MARKINGS AND PLACARDS ON THIS AIRPLANE APPLY TO ITS OPERATION AS A UTILITY CATEGORY AIRPLANE. FOR NORMAL AND UTILITY CATEGORY OPERATIONS, REFER TO THE AIRPLANE FLIGHT MANUAL.

FOR SPIN RECOVERY, USE FULL RUDDER AGAINST SPIN, FOLLOWED IMMEDIATELY BY FORWARD WHEEL.

NO ACROBATIC MANEUVERS (INCLUDING SPINS) ARE APPROVED FOR NORMAL CATEGORY OPERATIONS."

2. Adjacent to upper door latch:

"ENGAGE LATCH BEFORE FLIGHT."

3. On the inside of the baggage compartment door:

"MAXIMUM BAGGAGE 125 LBS." (S/N 671 to 1760A)  
(MAXIMUM BAGGAGE MAY BE INCREASED TO 200 LBS. IN ACCORDANCE WITH PIPER SERVICE SPARES LETTER NO. 242)

UTILITY CATEGORY OPERATION - NO BAGGAGE OR AFT PASSENGERS ALLOWED. NORMAL CATEGORY OPERATION - SEE AIRPLANE FLIGHT MANUAL WEIGHT AND BALANCE SECTION FOR BAGGAGE AND AFT PASSENGER LIMITATIONS.

4. In full view of the pilot:

"ROUGH AIR OR MANEUVERING SPEED 129 MPH." 112 KT

"UTILITY CATEGORY OPERATION - NO AFT PASSENGERS ALLOWED."

5. On the instrument panel in full view of the pilot when the oil cooler winterization kit is installed:

"OIL COOLER WINTERIZATION PLATE TO BE REMOVED WHEN AMBIENT TEMPERATURE EXCEEDS 50° F."

6. On the instrument panel in full view of the pilot when the autoflite is installed:

"FOR HEADING CHANGES: PRESS DISENGAGE SWITCH ON CONTROL WHEEL. CHANGE HEADING, RELEASE DISENGAGE SWITCH.

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Placards (Cont'd) 7. In full view of the pilot: "UTILITY CATEGORY ONLY."  
Acrobatic maneuvers are limited to the following:

		<u>Entry Speed</u>	
		Spins (Flaps Up).....	Stall
		Steep Turns.....	129 mph 112 KT
		Lazy Eights.....	129 112
		Chandelles.....	129 112
Airspeed	RED radial line	Never exceed	171 mph (148 knots)
Instrument	YELLOW arc	Caution Range (Smooth Air Only)	140 to 171 mph (121 to 148 knots).
Markings	GREEN arc	Normal Operating Range	67 to 140 mph (58 to 121 knots)
	WHITE arc	Flap Down Range	57 to 115 mph (50 to 100 knots)

2. Procedures  
Section

1. The stall-warning system is inoperative with the master switch off.
2. Electric fuel pump must be on for both landing and takeoff.
3. The PA-28-180 airplane is approved under FAA Regulation CAR 3 which prohibits intentional spins for normal category operation. The following information is noteworthy:
  - a. The stall characteristics of the PA-28-180 are normal with the nose pitching down moderately following the stall, occasionally with a moderate roll which can be corrected by normal use of ailerons and rudder against the roll.
  - b. Prolonged use of full rudder during stall practice may result in a rapid roll followed by a spin and should be avoided. Recovery from an incipient spin may be effected in less than one additional turn by use of opposite rudder followed by full forward control wheel.
  - c. In the event that a fully developed spin is inadvertently experienced, recovery is best made by using full opposite rudder followed by full forward wheel and full opposite aileron. The control positions against the spin should be maintained during the entire recovery, which may require several turns and a substantial loss of altitude if the airplane is loaded heavily with a rearward center of gravity.
4. Except as noted above, all operating procedures for this airplane are normal.

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Airplane Flight Manual  
Model PA 28-180

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Procedures Section  
(Cont'd.)

5. (Electric Pitch Trim Installation Only)  
The following emergency information applies in case of electric pitch trim malfunction:
- a. In case of malfunction, disengage electric pitch trim by pulling out circuit breaker on instrument panel.
  - b. In emergency, electric pitch trim may be overpowered using manual pitch trim.
  - c. In cruise configuration, malfunction results in 10° pitch change and 30 Ft. altitude variation.
6. (Autoflite Installation Only)  
The following emergency information applies in case of autoflite malfunction:
- a. In case of malfunction PRESS disconnect switch on pilot's control wheel.
  - b. Rocker switch on instrument panel - OFF.
  - c. Unit may be overpowered manually.
  - d. In cruise configuration malfunction, 3 seconds delay results in 60° bank, and 100 Ft. altitude loss.
  - e. In approach configuration malfunction, 1 second delay results in 10° bank and 0 Ft. altitude loss.
7. (AutoControl III Installation Only)
- I. Limitations:  
Pilot off during take off and landing.
  - II. Procedures:
    - a. Normal Operation  
Refers to Manufacturer's Operation Manual.
    - b. Emergency
      1. In case of malfunction, disengage manual controls.
      2. In emergency, pilot may be overpowered manually.
      3. In cruise configuration malfunction, 3 seconds delay results in 60° bank and 100 Ft. altitude loss.
      4. In approach configuration malfunction, 1 second delay results in 10° bank and 0 Ft. altitude loss.

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### 3. Performance Section

The following performance figures were obtained during FAA Type tests and may be realized under conditions indicated with the airplane and engine in good condition and with average piloting technique. All performance is given for 2400 pounds.

Loss of altitude during stalls varied from 125 to 200 feet, depending on configuration and power.

Stalling speeds, in mph, power off, versus angle of bank (Calibrated Airspeed):

Angle of bank	0	20	40	50	60
Flaps Up	67	69	76	83	94
Flaps Down	57	--	--	--	--

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EQUIPMENT LIST

MODEL PA-28-180

SERIAL NOS. 671 THRU 4377



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REVISION NO.	PAGE	DESCRIPTION	APPROVED	DATE
1	Title	Changed applicable Serial Nos. from 1 thru 4377 to 1 thru 5600.	<i>G. McLean</i>	7/15/69
2	Title	Changed applicable Serial Nos. from 1 thru 5600 to 671 thru 5600.	<i>G. McLean</i>	9/23/69
3	Title	Changed applicable Serial Nos. from 671 thru 5600 to 671 thru 4377.	<i>G. McLean</i>	5/8/69

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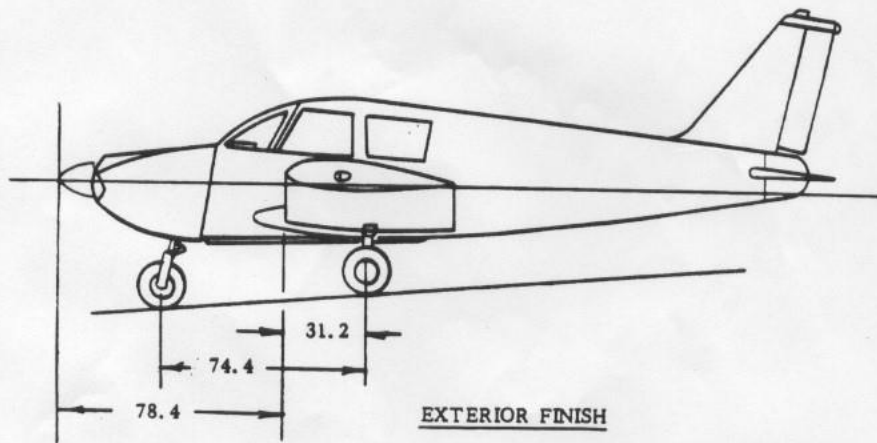
ACTUAL WEIGHT AND BALANCE

MODEL PA-28-180

SERIAL NUMBER 28 - \_\_\_\_\_

CERTIFICATE NUMBER \_\_\_\_\_

DATE \_\_\_\_\_



EXTERIOR FINISH

Base Color \_\_\_\_\_

1st Trim Color \_\_\_\_\_

2nd Trim Color \_\_\_\_\_

Registration No. Color \_\_\_\_\_

Type Finish \_\_\_\_\_

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WEIGHT AND BALANCE  
STANDARD EQUIPMENT LIST  
MODEL PA-28-180

Check if Installed	ITEM	WEIGHT (LBS)	ARM AFT DATUM (INCHES)	MOMENT (POUND- INCHES)
	<u>Engine Accessories</u>			
_____	Engine - Lycoming Model 0-360-A3A	274.4	26.1	6962
_____	Engine - Lycoming Model 0-360-A4A	282.4	26.1	7371
_____	Fuel Pump, Electric Auxiliary, Bendix Model 478360	1.8	41.8	75
_____	Fuel Pump, Engine Driven, Lycoming Drawing Nos. 73297, 74082, 75148 or 75246	1.6	41.3	66
_____	Oil Cooler, Piper Drawing, Harrison #C-8526250	2.6	18.1	47
_____	Filter, Fram Model CA-161 PL or AC No. A48C or Purolator AFP-2	.9	20.1	18
_____	Alternator, 35-amp, Chrysler No. 2098615	12.5	19.0	238
_____	Alternator, 60-amp, Chrysler No. 2642210 or 2642997	12.5	19.0	238
_____	Starter - Lycoming 74092 (Delco-Remy 1109511) *	18.0	19.5	351
_____	Starter - Lycoming 76211 (Prestolite MZ 4206) *	18.0	19.5	351
	<u>Propeller and Propeller Accessories</u>			
_____	Propeller, Sensenich M76EMM	34.5	10.1	348
_____	Propeller, Sensenich M76EMMS60	38.5	8.8	339
_____	Spinner and Attachment Plates	2.0	8.0	16

\* Included in Engine Weight.

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APPROVED				
Check if Installed	ITEM	WEIGHT (LBS)	ARM AFT DATUM (INCHES)	MOMENT (POUND- INCHES)
	<u>Landing Gear and Brakes</u>			
_____	Two Main Wheel Assemblies 6.00-6	32.0	109.6	3507
	(a) Cleveland Aircraft Products Wheel Assembly No. 40-28 Brake Assembly No. 30-18			
	(b) Two Main 4-Ply Rating Tires 6.00-6 with Regular Tubes			
_____	Two Main Wheel Assemblies	32.3	109.6	3540
	(a) Cleveland Aircraft Products Wheel Assembly No. 40-86 Brake Assembly No. 30-55			
	(b) Two Main 4-Ply Rating Tires 6.00-6 with Regular Tubes			
_____	One Nose Wheel 6.00-6	14.0	34.3	480
	(a) Cleveland Aircraft Products Wheel Assembly No. 38501 (Less Brake Drum)			
	(b) One Nose Wheel 4-Ply Rating Tire 6.00-6 with Regular Tubes			
	<u>Electrical Equipment</u>			
_____	Stall Warning Device, Safe Flight Instrument Corporation No. C52207-4	.2	80.2	16
_____	Voltage Regulator, Delco-Remy #118704	1.5	168.5	253
_____	Voltage Regulator, Chrysler #2098613	.5	57.8	29
_____	Voltage Regulator, Wico Electric #X-16300	.5	57.8	29
_____	Battery 12V, 25 A. H., Rebat Model S-25	21.5	160.9	3540

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Check if Installed	ITEM	WEIGHT (LBS.)	ARM AFT DATUM (INCHES)	MOMENT (POUND- INCHES)
	<u>Instrument</u>			
_____	Compass - Airpath No. C2350-L41	.9	66.6	60
_____	Airspeed Indicator, PAC 63205-2	.6	67.7	41
_____	Tachometer, AC 1548302	.8	67.7	54
_____	Tachometer, Stewart Warner PAC 62177-2 or 62177-3	.7	67.7	47
_____	Altimeter, Aero Marine No. 522	1.4	66.8	94
_____	Engine Cluster, PAC 63922-2	.8	68.8	55
_____	Engine Cluster, PAC 63426	.8	68.8	55
_____	Engine Cluster, PAC 63426-2	.8	68.8	55
	<u>Miscellaneous</u>			
_____	Fwd. Seat Belts	1.0	86.9	87
_____	Aft Seat Belts	.8	123.0	98
_____	Flight Manual	----	----	----
_____	Tow Bar	1.3	122.3	139
	<b>TOTAL</b>			
	AIRCRAFT EMPTY WEIGHT AS	_____	_____	_____
	(INCLUDES ITEMS CHECKED ON STANDARD EQUIPMENT LIST, UNUSABLE FUEL AND UNDRAINABLE OIL)			



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OPTIONAL EQUIPMENT LIST

MODEL PA-28-180

Check if Installed	ITEM	WEIGHT (LBS)	ARM AFT DATUM (INCHES)	MOMENT (POUND- INCHES)
	<u>Engine Accessories</u>			
_____	Vacuum Pump, Airborne Mechanisms Model No. 10-113A1, 113A5 or 200 cc and Drive	5.0	37.0	185
_____	Oil Filter - Lycoming #74911 (AC 81-A #6437032)	3.3	40.5	134
_____	Vacuum Regulator and Filter	2.2	57.0	125
	<u>Electrical Equipment</u>			
_____	Rotating Beacon, Grimes Model D7080	2.0	263.4	527
_____	Landing Light, G. E. Model 4509	.5	18.1	9
_____	Navigation Light (Rear) (1) Grimes Model 2064 (White)	.2	281.0	56
_____	Navigation Lights (2) Grimes Model A1285 (Red and Green)	.4	106.6	43
_____	Battery 12V, 35 A. H., Reading R-35	27.0	160.9	4344
_____	Cabin Light	.3	104.0	31
_____	Cabin Speaker	.8	104.0	83
_____	Rotating Beacon, Whelen Model WRM L-12	1.6	264.0	422

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Check if Installed	ITEM	WEIGHT (LBS)	ARM AFT DATUM (INCHES)	MOMENT (POUND- INCHES)
	<u>Electrical Equipment (Cont'd)</u>			
	Auxiliary Power Receptacle PAC 62225	2.7	168.0	454
	External Power Cable PAC 62355-2	4.6	142.8	657
	Piper Pitch Trim	4.0	158.0	632
	Heated Pitot Head	.4	100.0	40
	<u>Instruments</u>			
	Turn and Bank, Pioneer A-5	1.5	66.4	100
	Turn and Bank, Electric	2.7	65.8	178
	Suction Gauge, AN5771-11	.5	68.1	34
	Suction Gauge, Airborne Mechanisms 1G3-4	.5	68.1	34
	Suction Gauge, U. S. Gauge AW1821AFO3	.5	68.1	34
	Altimeter, AN5760-2 (C-12 or C-13)	Same as Standard Equipment Weight		
	Rate of Climb, Pioneer C-7	1.0	66.8	67
	Rate of Climb, AN5825	1.0	66.8	67
	Directional Gyro, Jack & Heintz	2.6	66.6	173
	Directional Gyro, Sperry	3.9	66.6	260
	Directional Gyro, Garwin (3")	2.4	65.6	157
	Directional Gyro, AIM (3")	3.1	64.9	201

PREPARED	PIPER AIRCRAFT CORP. DEVELOPMENT CENTER, VERO BEACH, FLA.	Weight and Balance Data Model PA-28-180
CHECKED		
APPROVED		REPORT VB-164

Check if Installed	ITEM	WEIGHT (LBS)	ARM AFT DATUM (INCHES)	MOMENT (POUND- INCHES)
	<u>Instruments</u> (Cont'd)			
_____	Artificial Horizon, Jack & Heintz	2.8	66.1	185
_____	Artificial Horizon, Garwin (3")	1.8	65.8	118
_____	Artificial Horizon, AIM (3")	2.2	65.3	144
_____	Air Temperature Gauge, Rochester Manufacturing Co., No. 1592-C2 or NHM-70 (Manning, Maxwell & Moore)	.2	82.6	17
_____	Clock, 8-Day, MIL-C-7939	.4	68.3	27
_____	Tru-Speed Indicator, PAC 62143-2	Same as Standard Equipment Weight		
_____	Piper Course Selector PAC 31058	3.0	66.6	200
_____	Electric Turn and Bank	2.7	65.8	178
_____	Pictorial Rate of Turn, Mitchell 52D69	1.3	66.2	86
_____	Rate of Climb, Karnish AC135-3	1.0	66.8	67
_____	Brittain Turn Coordinator #TC-100(12)	2.6	65.6	171
	<u>AutoPilots</u>			
	<u>AutoControl II</u>			
_____	Roll Servo, Mitchell #1X221E-CH-1	2.8	60.6	170
_____	Console, Mitchell #1X224E-3	1.3	66.6	87
_____	Directional Gyro, Mitchell #S2B15E or	4.3	66.6	286
_____	Directional Gyro, Course Selector PAC Drawing 31058-2	3.0	66.6	200

PREPARED	PIPER AIRCRAFT CORP. DEVELOPMENT CENTER, VERO BEACH, FLA.	Weight and Balance Data Model PA-28-180
CHECKED		
APPROVED	REPORT VB-164	PAGE 8 Section 1

Check if Installed	ITEM	WEIGHT (LBS)	ARM AFT DATUM (INCHES)	MOMENT (POUND- INCHES)
	<u>AutoPilots</u> (Cont'd)			
_____	Artificial Horizon, Mitchell #52B9	4.5	66.1	298
_____	AutoControl III			
_____	Roll Servo, Mitchell #1D363-183R	2.5	122.2	306
_____	Console, Mitchell #1C338	1.2	66.6	80
_____	Cables	.7	95.5	67
_____	Attitude Gyro, Mitchell #52D66 (Garwin)	1.9	65.8	125
_____	Attitude Gyro, Mitchell #52D66 (AIM)	2.3	65.3	150
_____	Directional Gyro, Mitchell #52D54P (Garwin)	2.5	65.6	164
_____	Directional Gyro, Mitchell #52D54P (AIM)	3.2	64.9	208
_____	Omni Coupler	.9	65.8	59
_____	AutoFlite			
_____	Roll Servo, Mitchell #1D363-153	2.6	122.2	318
_____	Gyro Amplifier, Mitchell #1C359	1.8	111.8	201
_____	Cables	1.0	95.5	96
_____	Panel Unit	.3	68.8	21
_____	Omni Tracker (#1D482)	.5	64.5	32

PREPARED	PIPER AIRCRAFT CORP. DEVELOPMENT CENTER, VERO BEACH, FLA.	Weight and Balance Data Model PA-28-180
CHECKED		
APPROVED	REPORT VB-164	PAGE 9 Section 1

Check if Installed	ITEM	WEIGHT (LBS)	ARM AFT DATUM (INCHES)	MOMENT (POUND- INCHES)
<input type="checkbox"/>	<u>Radio</u>			
<input type="checkbox"/>	PM-1 Marker Beacon			
<input type="checkbox"/>	Receiver	1.1	121.3	133
<input type="checkbox"/>	Panel Unit	.3	69.0	21
<input type="checkbox"/>	Cable	.3	85.0	26
<input type="checkbox"/>	Piper Radio Compass PRC-3	4.5	64.4	290
<input type="checkbox"/>	Piper VHF Transceiver PTR-1	5.0	64.8	324
<input type="checkbox"/>	Piper Omni Convertor O-1	2.5	65.3	163
<input type="checkbox"/>	King KX150B	9.1	62.8	572
<input type="checkbox"/>	Omni Receiving Antenna, Narco VTP-37 (Includes Cables)	1.4	203.0	284
<input type="checkbox"/>	VHF Antenna, Transmitting VHF-1	.3	157.8	47
<input type="checkbox"/>	VHF Antenna, Transmitting VHF-2	.3	192.8	58
<input type="checkbox"/>	Cable, VHF-1	.4	118.0	47
<input type="checkbox"/>	Cable, VHF-2	.5	135.0	68
<input type="checkbox"/>	Low Frequency Antenna	.5	167.0	84
<input type="checkbox"/>	Loop Antenna (PRC-3)	.3	54.5	16
<input type="checkbox"/>	Narco Mark 12A			
<input type="checkbox"/>	Transceiver, Single	6.0	62.8	377
<input type="checkbox"/>	Transceiver, Dual	12.0	62.8	754
<input type="checkbox"/>	Modulator-Power Unit, Single	4.0	56.0	224
<input type="checkbox"/>	Modulator-Power Unit, Dual	8.0	186.0	1488



PREPARED		PIPER AIRCRAFT CORP. DEVELOPMENT CENTER, VERO BEACH, FLA.	Weight and Balance Data Model PA-28-180	
CHECKED				
APPROVED		REPORT VB-164	PAGE 10 Section 1	
Check if Installed	ITEM	WEIGHT (LBS)	ARM AFT DATUM (INCHES)	MOMENT (POUND- INCHES)
	<u>Radio</u> (Cont'd)			
_____	Cable, Single	.3	58.0	17
_____	Cable, Dual	3.4	120.0	408
_____	Narco VOA-6 Omni Convertor	1.8	65.3	118
_____	Narco VOA-5 Omni Convertor	3.1	65.3	202
_____	Narco VOA-4 Omni Convertor	3.0	65.3	196
_____	Narco ADF-30	9.9	107.9	1068
_____	Narco Omnigator VTR-2A Installation (Less Antenna)	14.0	58.0	812
_____	Marker Antenna	1.2	64.8	78
_____	Piper Radio Compass PRC-4	4.9	64.4	316
_____	Loop Antenna (PRC-4)	.4	112.6	45
_____	Piper Omni Convertor OL-1	2.8	65.3	183
_____	Narco ADF-31			
_____	Receiver	5.1	64.4	328
_____	Loop Antenna	2.7	162.0	437
_____	Cable Antenna	1.7	108.0	184
_____	Bendix ADF-T-12C			
_____	Receiver	3.8	64.9	247
_____	Audio Amplifier	.8	64.9	52
_____	Radio Compass	1.7	67.3	114

PREPARED	PIPER AIRCRAFT CORP. DEVELOPMENT CENTER, VERO BEACH, FLA.	Weight and Balance Data Model PA-28-180
CHECKED		
APPROVED		REPORT VB-164

Check if Installed	ITEM	WEIGHT (LBS)	ARM AFT DATUM (INCHES)	MOMENT (POUND- INCHES)
	<u>Radio</u> (Cont'd)			
_____	Loop Antenna	1.2	160.8	193
_____	Cable, Antenna	1.5	108.0	162
_____	Narco - UDI-III DME	8.6	62.6	538
_____	Narco Mark III	7.5	63.6	477
_____	Narco UDI-4 DME			
_____	Receiver	8.5	62.6	532
_____	Antenna	.3	113.9	34
_____	Cable, Antenna	.4	100.0	40
_____	UGR-2 Glide Slope			
_____	Receiver	2.4	173.8	417
_____	Cable	2.1	128.0	269
_____	Antenna	.4	92.4	37
_____	Cable, Antenna	.5	145.0	73
_____	Transmitter Selector (Dual VHF Only)	.7	67.2	47
_____	Microphone	.5	75.0	38
_____	Headset	.5	66.0	33
_____	Junction Box	.6	67.2	40

PREPARED	PIPER AIRCRAFT CORP. DEVELOPMENT CENTER, VERO BEACH, FLA.	Weight and Balance Data Model PA-28-180
CHECKED		
APPROVED	REPORT VB-154	PAGE 12 Section 1

Check if Installed	ITEM	WEIGHT (LBS)	ARM AFT DATUM (INCHES)	MOMENT (POUND- INCHES)
	<u>Miscellaneous</u>			
_____	Nose Wheel Fairing	3.5	34.8	122
_____	Main Wheel Fairing	7.4	109.6	811
_____	Assist Step	1.8	156.0	281
_____	Toe Brakes (Dual)	10.5	54.6	573
_____	Toe Brakes (Single)	5.0	54.6	273
_____	Fire Extinguisher-Stop Fire #A-20	7.5	93.0	698
_____	Inertia Safety Belt PAC 65766	2.5	111.6	279
_____	Assist Strap and Coar Hooks	.2	109.5	22
_____	Lighter	.2	68.8	14
_____	Fire Extinguisher, Kidde Kompact VI (With Brackets)	5.3	85.0	451

TOTAL

EMPTY C. G. AFT DATUM IS \_\_\_\_\_

AIRCRAFT EMPTY WEIGHT \_\_\_\_\_

OPTIONAL EQUIPMENT WEIGHT \_\_\_\_\_

LICENSED EMPTY WEIGHT \_\_\_\_\_

PREPARED	PIPER AIRCRAFT CORP. DEVELOPMENT CENTER, VERO BEACH, FLA.	Weight and Balance Data Model PA-28-180
CHECKED		
APPROVED		REPORT VB-164

IT IS THE RESPONSIBILITY OF THE PILOT AND AIRCRAFT OWNER TO INSURE THAT THE AIRPLANE IS LOADED PROPERLY. THE EMPTY WEIGHT C.G. IS FOR THE AIRPLANE AS DELIVERED FROM THE FACTORY. REFER TO FORM FAA-337 WHEN ALTERATIONS HAVE BEEN MADE.

C. G. RANGE AND WEIGHT INSTRUCTIONS

1. Add the weight of all items to be loaded to the licensed empty weight.
2. Use the loading graph to determine the moment of all items to be carried in the airplane.
3. Add the moment of all items to be loaded to the licensed empty weight moment.
4. Divide the total weight moment by the total weight to determine the C. G. location.
5. By using the figures of item 1 and item 4, locate a point on the C. G. range and weight graph. If the point falls within the C. G. envelope, the loading meets all weight and balance requirements.

SAMPLE LOADING PROBLEM (NORMAL CATEGORY)

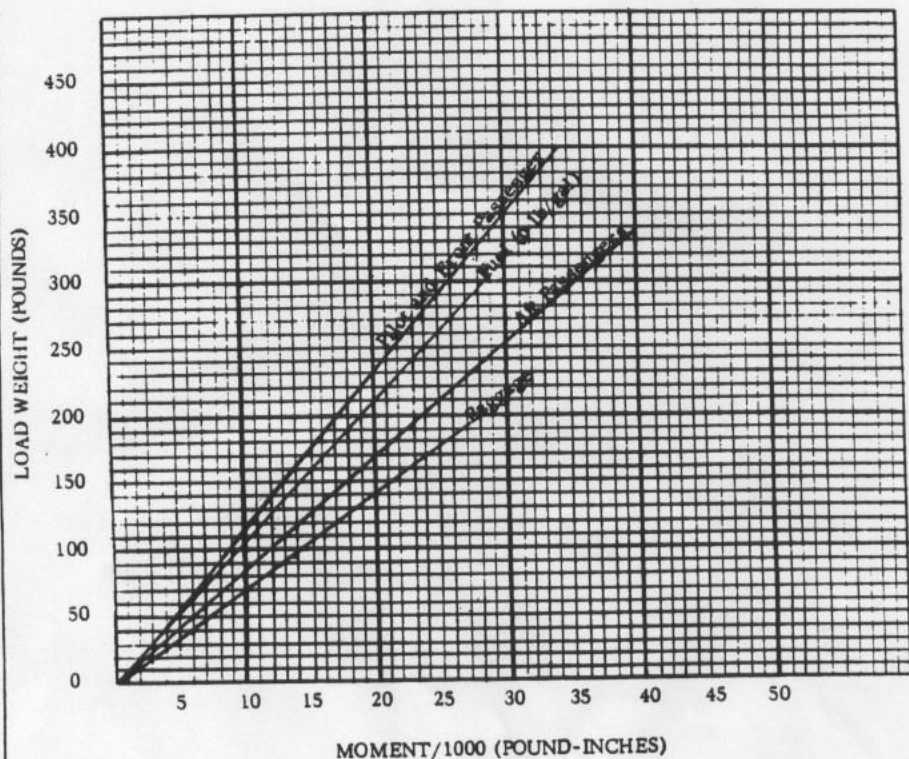
	<u>WEIGHT (LBS)</u>	<u>ARM AFT DATUM (INCHES)</u>	<u>MOMENT (POUND-INCHES)</u>
LICENSED EMPTY WEIGHT			
OIL (2 GALLON)	15	32.5	488
PILOT & PASSENGER	340	85.5	29070
FUEL		95.0	
PASSENGERS (REAR SEAT) *	340	118.1	40154
BAGGAGE *		142.8	
TOTAL LOADED AIRPLANE			
		=	
			INCHES (ARM AFT DATUM)

LOCATE THIS POINT ( ) ON THE C. G. RANGE AND WEIGHT GRAPH. SINCE THIS POINT FALLS WITHIN THE C. G. ENVELOPE THE LOADING MEETS ALL WEIGHT AND BALANCE REQUIREMENTS.

- \* Utility Category Operation - No baggage or aft passengers allowed.
- Normal Category Operation - Maximum baggage 125 lbs. (S/N 671 to 1760A).  
Maximum baggage 200 lbs. (S/N 1761 and up).  
Check aft C. G. between 150 lbs. and 200 lbs.

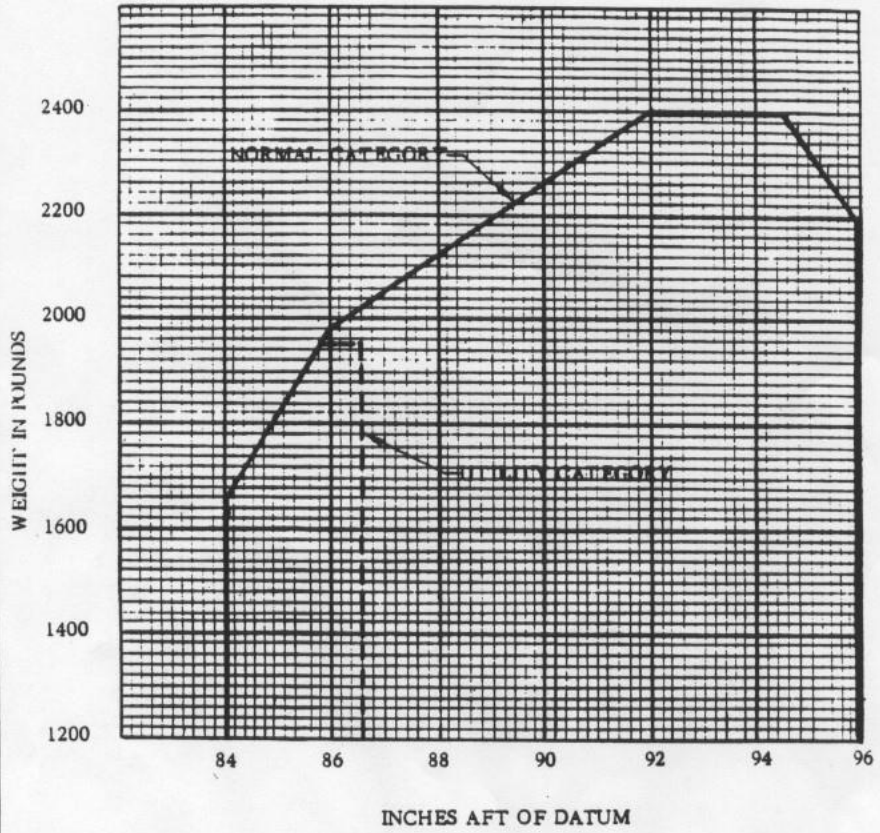
PREPARED	PIPER AIRCRAFT CORP. DEVELOPMENT CENTER, VERO BEACH, FLA.	Weight and Balance Data Model PA-28-180
CHECKED		
APPROVED	REPORT VB-164	PAGE 14 Section I

LOADING GRAPH





C. G. RANGE AND WEIGHT



The following pages are for reference and are not formally part of the aircraft flight manual for FZPX

### Summary of Speeds (knots)

V<sub>so</sub> - 50

V<sub>s</sub> - 58

V<sub>x</sub> - 64

V<sub>y</sub> - 74

V<sub>fe</sub> - 100

V<sub>a</sub> - 112

V<sub>no</sub> - 121

V<sub>ne</sub> - 148

Enroute climb - 87

Best Glide - 71

Max Crosswind - 17

Flaps up approach - 75

Full flap V<sub>ref</sub> - 65

Tire pressure - 24 psi

# CHEROKEE D

PA-28-180

## Owner's Handbook



**Piper Aircraft Corporation, Vero Beach, Florida  
U. S. A.**

REFERENCE ONLY

THIS ELECTRONIC VERSION  
OF THE AOH IS  
NOT APPROVED TO  
REPLACE ANY OPERATING  
INFORMATION REQUIRED  
BY THE REGULATIONS.

## NOTICE

THIS HANDBOOK IS NOT DESIGNED, NOR CAN ANY HANDBOOK SERVE, AS A SUBSTITUTE FOR ADEQUATE AND COMPETENT FLIGHT INSTRUCTION, OR KNOWLEDGE OF THE CURRENT AIRWORTHINESS DIRECTIVES, THE APPLICABLE FEDERAL AIR REGULATIONS, AND ADVISORY CIRCULARS. IT IS NOT INTENDED TO BE A GUIDE OF BASIC FLIGHT INSTRUCTION, NOR A TRAINING MANUAL.

THE HANDBOOK IS DESIGNED:

1. TO HELP YOU OPERATE YOUR CHEROKEE WITH SAFETY AND CONFIDENCE.
2. TO MORE FULLY ACQUAINT YOU WITH THE BASIC PERFORMANCE AND HANDLING CHARACTERISTICS OF THE AIRPLANE.
3. TO MORE FULLY EXPLAIN YOUR CHEROKEE'S OPERATION THAN IS PERMISSIBLE TO SET FORTH IN THE AIRPLANE FLIGHT MANUAL.

IF THERE IS ANY INCONSISTENCY BETWEEN THIS HANDBOOK AND THE AIRPLANE FLIGHT MANUAL APPROVED BY THE F.A.A., THE FLIGHT MANUAL SHALL GOVERN.

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Revised text and illustrations shall be indicated by a black vertical line in the margin opposite the change.

Additional copies of this manual, Part No. 753 765 may be obtained from your Piper Dealer.

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753 765  
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**SECTION I**  
**SPECIFICATIONS**

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## SECTION I

### SPECIFICATIONS

#### PERFORMANCE

Performance figures are for airplanes equipped for cross-country transportation and flown at gross weight under standard conditions at sea level or stated altitude. Any changes in equipment may result in changes in performance.

Take-off Run (maximum effort, 25° flap) (ft)	720	
Take-off over 50-ft barrier (maximum effort, 25° flap) (ft)	1625	
Best Rate of Climb Speed (mph)	85	74 KT
Rate of Climb (ft per min)	750	
Service Ceiling (ft)	13,000	
Absolute Ceiling (ft)	15,000	
Top Speed (mph)	152	132 KT
Optimum Cruising Speed (75% power, optimum altitude) (mph)	143	124 KT
Cruising Range (75% power, optimum altitude) (mi)	725	
Optimum Cruising Range (55% power, optimum altitude) (mi)	845	
Stalling Speed (flaps down) (mph)	57	50 KT
Stalling Speed (flaps up) (mph)	67	58 KT
Landing Roll (flaps down) (ft)	600	
Landing Roll over 50-ft barrier (ft)	1150	

**SPECIFICATIONS (cont):****WEIGHTS**

Gross Weight (lbs)	2400
Empty Weight (Standard) (lbs)	1294
USEFUL LOAD (Standard) (lbs)	1106

**POWER PLANT**

Engine (Lycoming)	O-360-A4A
Rated Horsepower	180
Rated Speed (rpm)	2700
Bore (in.)	5.125
Stroke (in.)	4.375
Displacement (cu in.)	361.0
Compression Ratio	8.5:1
Dry Weight (lbs)	285
Propeller	<del>M76EMMS60</del> 76EM8-0-60

**FUEL AND OIL**

Fuel Capacity (U.S. gal)	50
Oil Capacity (qts)	8
Fuel, Aviation Grade (min octane)	91/96

**BAGGAGE**

Maximum Baggage (lbs)	200
Baggage Space (cu ft)	17
Baggage Door Size (in.)	20 x 22

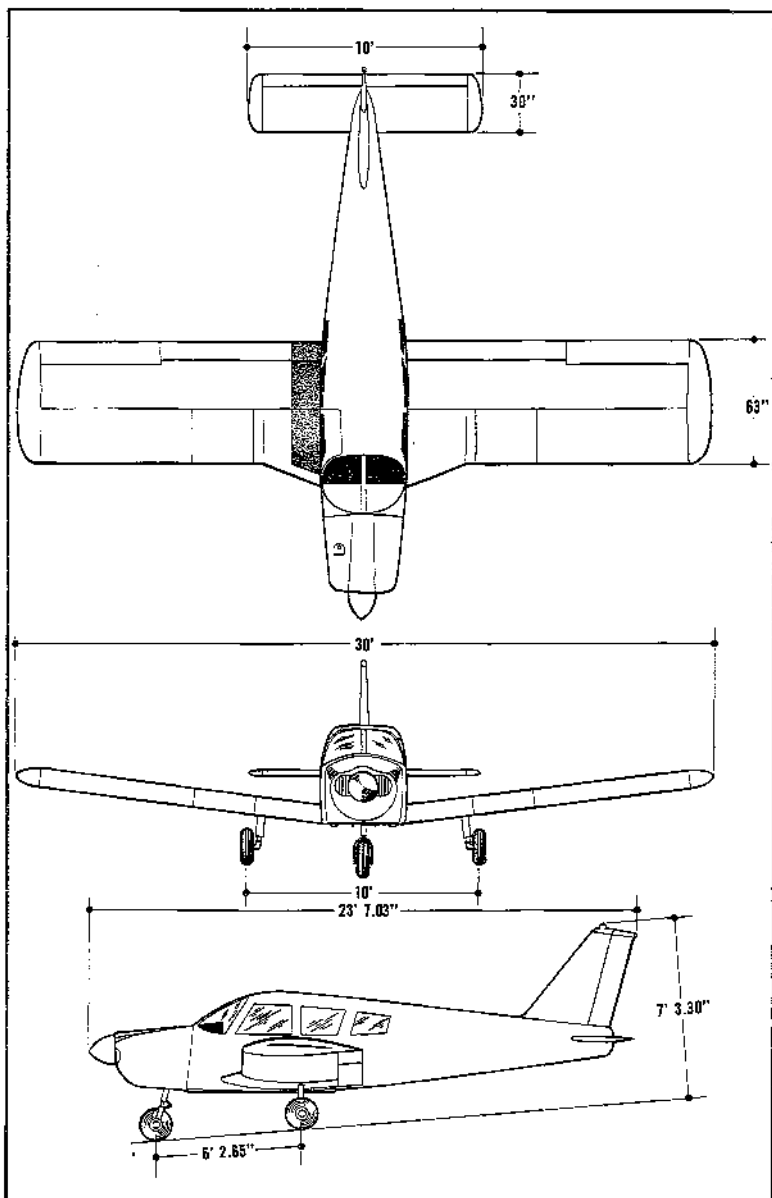
**SPECIFICATIONS (cont):**

**DIMENSIONS**

Wing Span (ft)	30
Wing Area (sq ft)	160
Wing Loading (lbs per sq ft)	15.0
Length (ft)	23.5
Height (ft)	7.3
Power Loading (lbs per hp)	13.3

**LANDING GEAR**

Wheel Base (ft)	6.2
Wheel Tread (ft)	10
Tire Pressure (psi) Nose	24
Main	24
Tire Size                   Nose (4 ply rating)	6.00 x 6
Main (4 ply rating)	6.00 x 6



**SECTION II**  
**DESIGN INFORMATION**

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## SECTION II

### DESIGN INFORMATION

#### ENGINE AND PROPELLER

The Cherokee "D" is powered by a Lycoming O-360-A4A four cylinder, direct drive, horizontally opposed engine rated at 180 HP at 2700 RPM. It is furnished with a starter, 60 ampere 12 volt alternator, shielded ignition, vacuum pump drive, fuel pump, and a dry, automotive type carburetor air filter.

The exhaust system is of the cross-over type to reduce back pressure and improve performance. It is made entirely from stainless steel and is equipped with dual mufflers. A heater shroud around the mufflers is provided to supply heat for the cabin and windshield defrosting. 76EM8-0-60

The Sensenich ~~M76EMMS60~~ fixed-pitch propeller is made from a one-piece alloy forging.

#### STRUCTURES

All structures are of aluminum alloy construction and are designed to ultimate load factors well in excess of normal requirements. All exterior surfaces are primed with etching primer and painted with acrylic enamel.

The wings are attached to each side of the fuselage by inserting the buttends of the respective main spars into a spar box carry-through which is an integral part of the fuselage structure, providing in effect a continuous main spar with splices at each side of the fuselage. There are also fore and aft attachments at



the rear spar and at an auxiliary front spar.

The wing airfoil section is a laminar flow type, NACA65<sub>2</sub>-415 with the maximum thickness about 40% aft of the leading edge. This permits the main spar carry-through structure to be located under the rear seat providing unobstructed cabin floor space ahead of the rear seat.

## LANDING GEAR

The three landing gears use a Cleveland 6.00 x 6 wheel, the main wheels being provided with Cleveland single disc hydraulic brake assemblies, No. 30-55. All wheels use 6.00 x 6 four ply tires with tubes.

The nose gear is steerable through a 44 degree arc by use of the rudder pedals. A spring device is incorporated in the rudder pedal torque tube assembly to aid in rudder centering and to provide rudder trim. The nose gear steering mechanism also incorporates a hydraulic shimmy dampener.

The three struts are of the air-oil type, with the normal extension being 3.25 inches for the nose gear and 4.50 inches for the main gear.

The standard brake system for the Cherokee consists of a hand lever and master cylinder which is located below and behind the left center of the instrument sub-panel. The brake fluid reservoir is installed on the top left front face of the firewall. The parking brake is incorporated in the master cylinder and is actuated by pulling back on the brake lever, depressing the knob attached to the handle and releasing the brake lever. To release the parking brake, pull back on the lever to disengage the catch mechanism and allow the handle to swing forward.

Optional toe brakes are available to supplement the standard hand lever and parking brake system.

## CONTROL SYSTEMS

Dual controls are provided as standard equipment with a cable system used between the controls and the surfaces. The horizontal tail is of the Flying Tail type (stabilator), with a trim tab mounted on the trailing edge of the stabilator to reduce the control system forces. This tab is actuated by a control wheel on the floor between the front seats. The stabilator provides extra stability and controllability with less size, drag and weight than conventional tail surfaces. The ailerons are provided with a differential action which tends to reduce adverse yaw in turning maneuvers, and which also reduces the amount of coordination required in normal turns. A rudder trim adjustment is mounted on the right side of the pedestal below the throttle quadrant and permits directional trim as needed in flight.

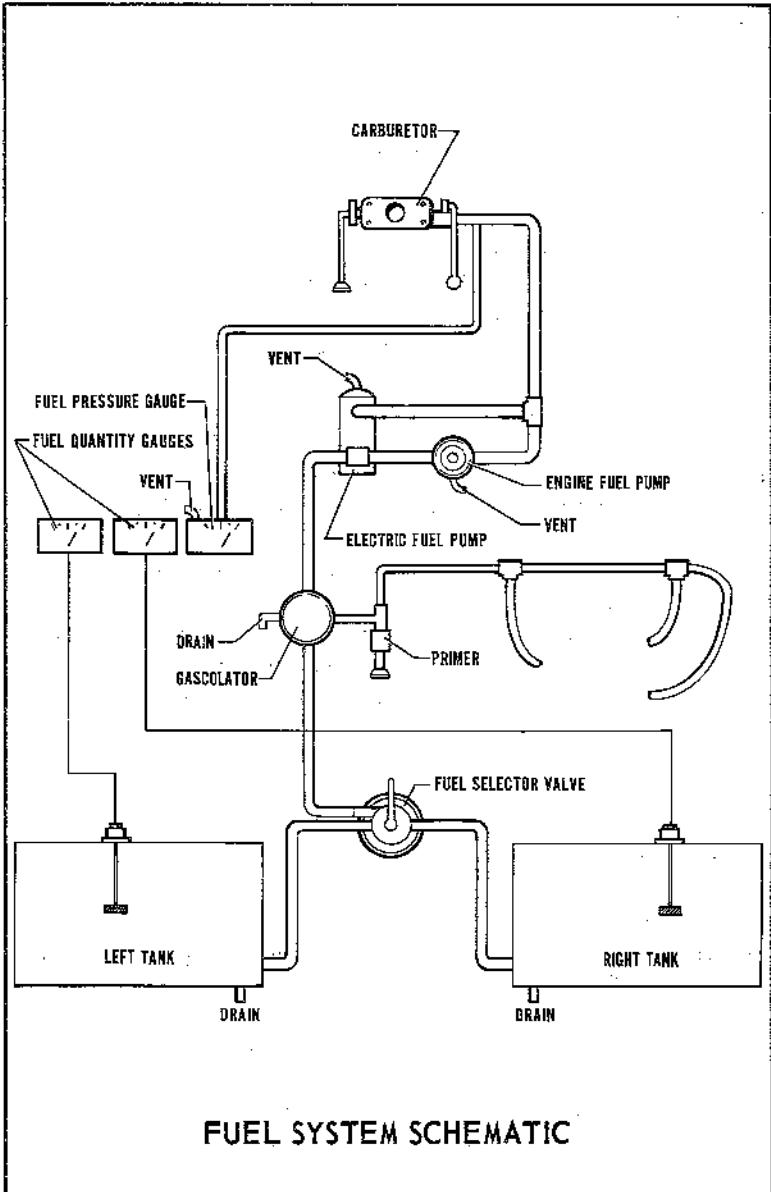
The flaps are manually operated, balanced for light operating forces and spring-loaded to return to the up position. A past-center lock incorporated in the actuating linkage holds the flap when it is in the up position so that it may be used as a step on the right side. The flap will not support a step load except when in the full up position, so it must be completely retracted when used as a step. The flaps have three extended positions: 10, 25 and 40 degrees.

## FUEL SYSTEM

Fuel is stored in two twenty-five gallon tanks which are secured to the leading edge structure of each wing by screws and nut plates. This allows easy removal for service or inspection.

The fuel selector control is located on the left side-panel, forward of the pilot's seat. If a modified selector valve cover has been installed, the button on the selector cover must be depressed and held while the handle is moved to the OFF position. The button releases automatically when the handle is moved into the ON position.

An auxiliary electric fuel pump is provided in case of failure



FUEL SYSTEM SCHEMATIC

of the engine-driven pump. The electric pump should be on for all take-offs and landings, and when switching tanks. The pump switch is located in the switch panel above the throttle quadrant.

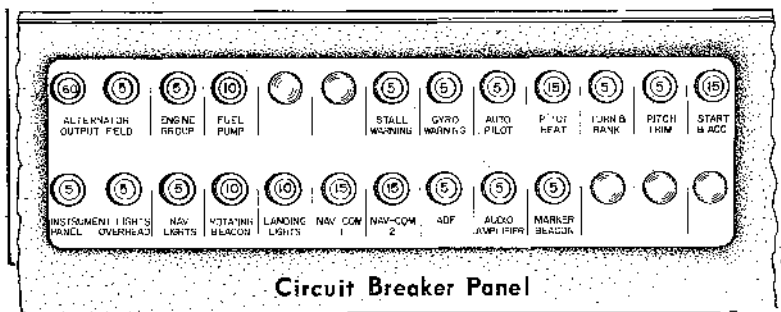
Each tank has an individual quick drain located at the bottom, inboard rear corner, and should be drained to check for water before each flight. The fuel strainer, which is also equipped with a quick drain, is located on the front lower left corner of the firewall. This strainer should be drained regularly to check for water or sediment accumulation.

Fuel quantity and pressure are indicated on gauges located in a cluster on the left side of the instrument panel.

**ELECTRICAL SYSTEM**

The electrical system includes a 12 volt 60 amp alternator, battery, voltage regulator, overvoltage relay and master switch relay. The battery is mounted in a stainless steel box immediately aft of the baggage compartment. The regulator and overvoltage relay are located on the forward left side of the fuselage behind the instrument panel.

Electrical switches are located on the right center instrument panel, and the circuit breakers are located on the lower right instrument panel. A rheostat switch on the left side of the switch panel controls the navigation lights and the dome instrument light. It also dims the dome light. The similar switch on the right side controls and dims the panel lights.



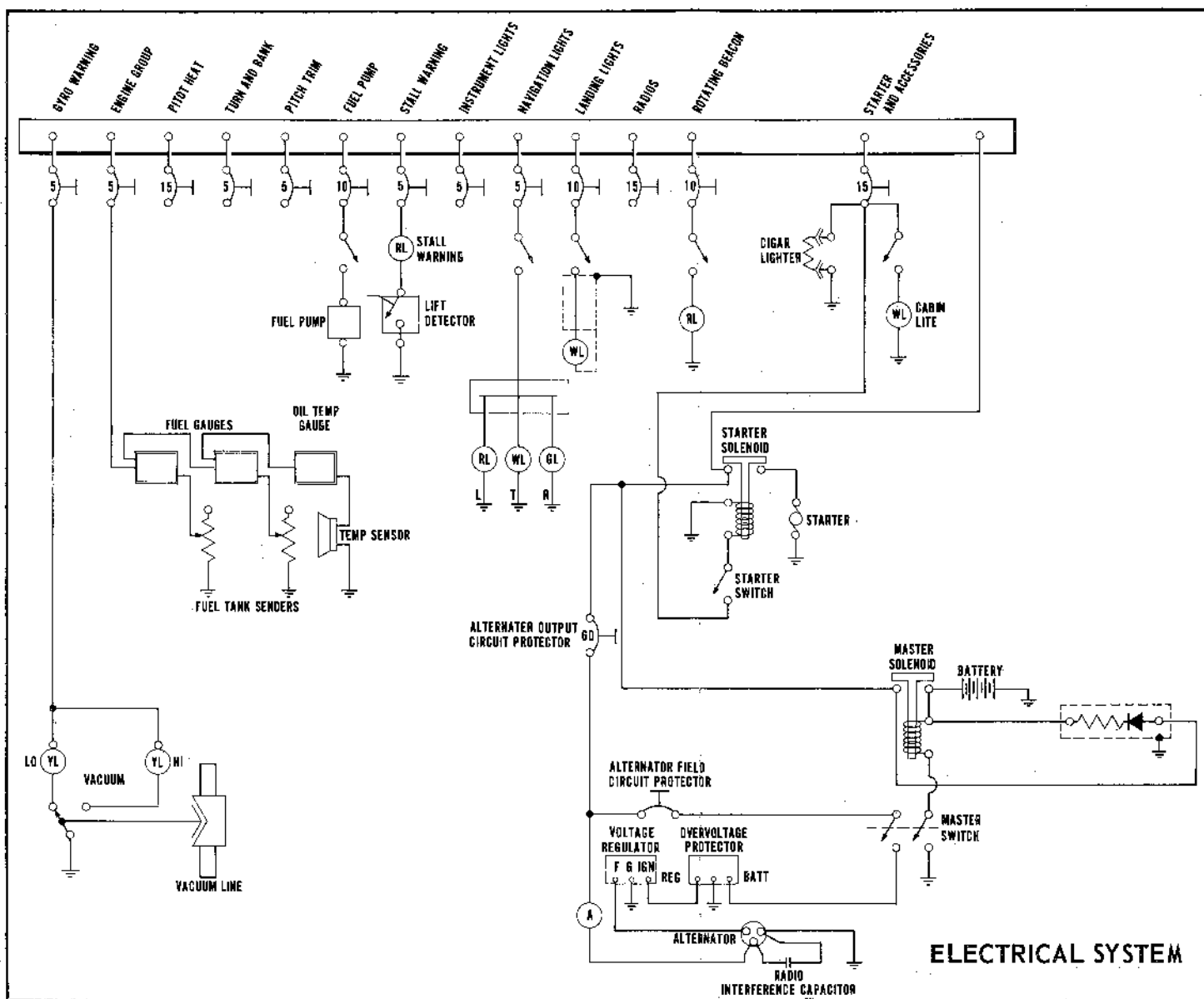
Standard accessories include a starter, electric fuel pump, stall warning indicator, cigar lighter, fuel gauge and ammeter. The navigation lights, anti-collision light, landing light, instrument lighting and cabin dome light are optional. Circuits will handle an entire complement of communications and navigational equipment.

The alternator system offers many advantages over the generator system both in operation and maintenance. The main advantage is full electrical power output at lower engine RPM. This is a great improvement for radio and electrical equipment operation. Since the alternator output is available at all times, the battery will be charging for a greater percentage of use. This will make cold-morning starting easier.

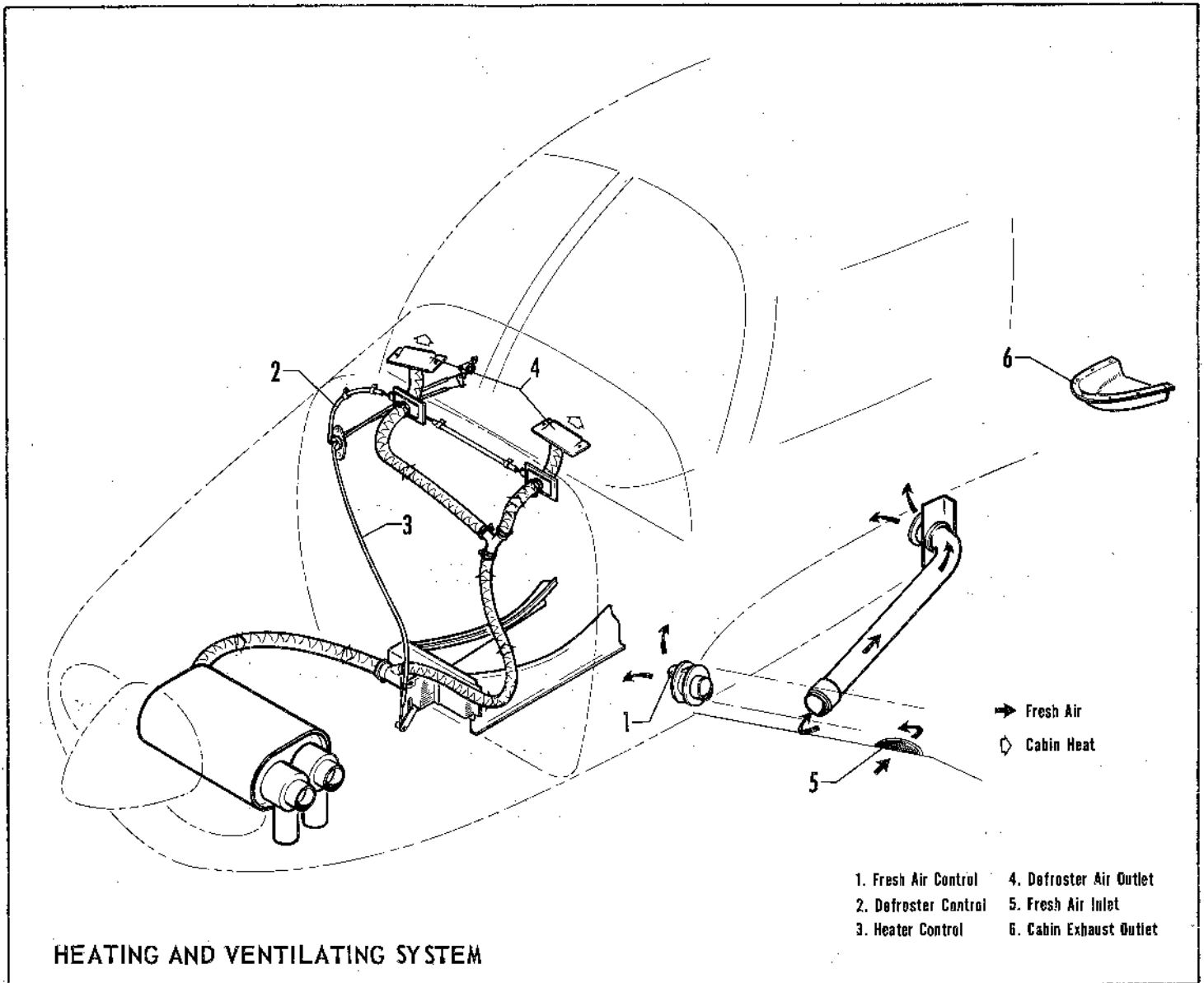
Unlike previous generator systems, the ammeter does not indicate battery discharge; rather it displays in amperes the load placed on the alternator. With all electrical equipment off (except master switch) the ammeter will be indicating the amount of charging current demanded by the battery. As each item of electrical equipment is turned on, the current will increase to a total appearing on the ammeter. This total includes the battery. The maximum continuous load for night flight, with radios on, is about 30 amperes. This 30 ampere value, plus approximately two amperes for a fully charged battery, will appear continuously under these flight conditions. The amount of current shown on the ammeter will tell immediately if the alternator system is operating normally, as the amount of current shown should equal the total amperage drawn by the equipment which is operating.

If no output is indicated on the ammeter during flight, reduce the electrical load by turning off all unnecessary electrical equipment. Check both 5 ampere field breaker and 60 ampere output breaker and reset if open. If neither circuit breaker is open, turn off the master switch for 30 seconds to reset the overvoltage relay. If ammeter continues to indicate no output, maintain minimum electrical load and terminate flight as soon as practical.

Maintenance on the alternator should prove to be a minor factor. Should service be required, contact the local Piper Dealer.







## HEATING AND VENTILATING SYSTEM

Heat for the cabin interior and the defroster system is provided by a heater muff attached to the exhaust system. The amount of heat desired can be regulated with the controls located on the far right side of the instrument panel.

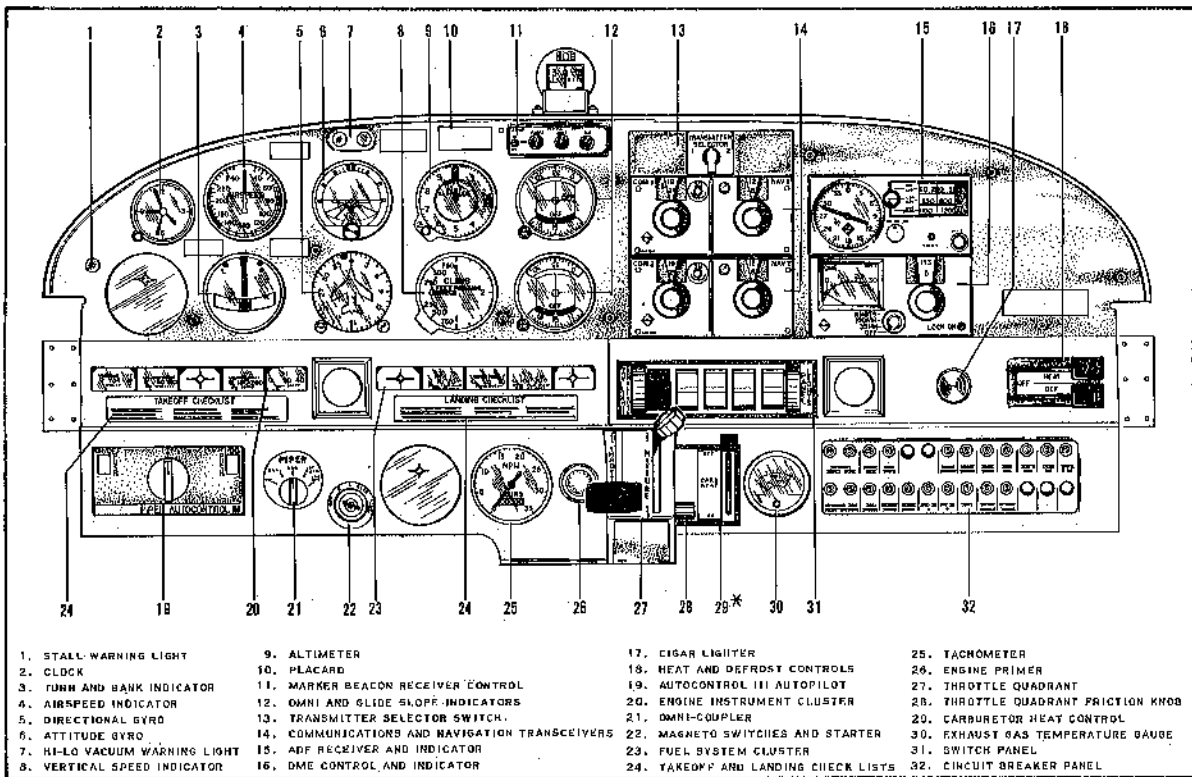
Fresh air inlets are located in the leading edge of the wing at the intersection of the tapered and straight sections. A large adjustable outlet is located on the side of the cabin near the floor at each seat location. Cabin air is exhausted through an outlet located below the rear seat.

## CABIN FEATURES

The instrument panel of the Cherokee is designed to accommodate the customary advanced flight instruments and the normally required power plant instruments. The Artificial Horizon and Directional Gyro are vacuum operated through use of a vacuum pump installed on the engine, while the Turn and Bank instrument is electrically operated. Above the Attitude Gyro are two lights which indicate high or low vacuum. A natural separation of the flight group and the power group is provided by placing the flight group in the upper instrument panel and the power group in the center and lower instrument panels. The radios and circuit breakers are on the right hand instrument panel, and extra circuits are provided for a complete line of optional radio equipment.

The cabin interior includes a pilot storm window, two sun visors, ash trays, two map pockets, and pockets on the backs of each front seat. The front seats are adjustable fore and aft for pilot-passenger comfort and ease of entry and exit. Recessed arm rests are also provided for the front seats.

The 17 cubic foot baggage area may be reached from the cabin or through a large 20 x 22 inch outside door.



- |                               |  |                                     |                                     |
|-------------------------------|--|-------------------------------------|-------------------------------------|
| 1. STALL WARNING LIGHT        | 9. ALTIMETER                                   | 17. CIGAR LIGHTER                   | 25. TACHOMETER                      |
| 2. CLOCK                      | 10. PLACARD                                    | 18. HEAT AND DEFROST CONTROLS       | 26. ENGINE PRIMER                   |
| 3. TURN AND BANK INDICATOR    | 11. MARKER BEACON RECEIVER CONTROL             | 19. AUTOCONTROL III AUTOPILOT       | 27. THROTTLE QUADRANT               |
| 4. AIRSPEED INDICATOR         | 12. OMNI AND GLIDE SLOPE INDICATORS            | 20. ENGINE INSTRUMENT CLUSTER       | 28. THROTTLE QUADRANT FRICTION KNOB |
| 5. DIRECTIONAL GYRO           | 13. TRANSMITTER SELECTOR SWITCH                | 21. OMNI-COUPLER                    | 29. CARBURETOR HEAT CONTROL         |
| 6. ATTITUDE GYRO              | 14. COMMUNICATIONS AND NAVIGATION TRANSCEIVERS | 22. MAGNETO SWITCHES AND STARTER    | 30. EXHAUST GAS TEMPERATURE GAUGE   |
| 7. HI-LO VACUUM WARNING LIGHT | 15. ADF RECEIVER AND INDICATOR                 | 23. FUEL SYSTEM CLUSTER             | 31. SWITCH PANEL                    |
| 8. VERTICAL SPEED INDICATOR   | 16. DME CONTROL AND INDICATOR                  | 24. TAKEOFF AND LANDING CHECK LISTS | 32. CIRCUIT BREAKER PANEL           |

**NOTE**

\*In aircraft with Serial Nos. 28-4378 to 28-5042 carburetor heat is ON when indicator is in the UP position. From Serial Nos. 28-5043 and up, carburetor heat is ON when indicator is in the DOWN position.

**SECTION III**  
**OPERATING INSTRUCTIONS**

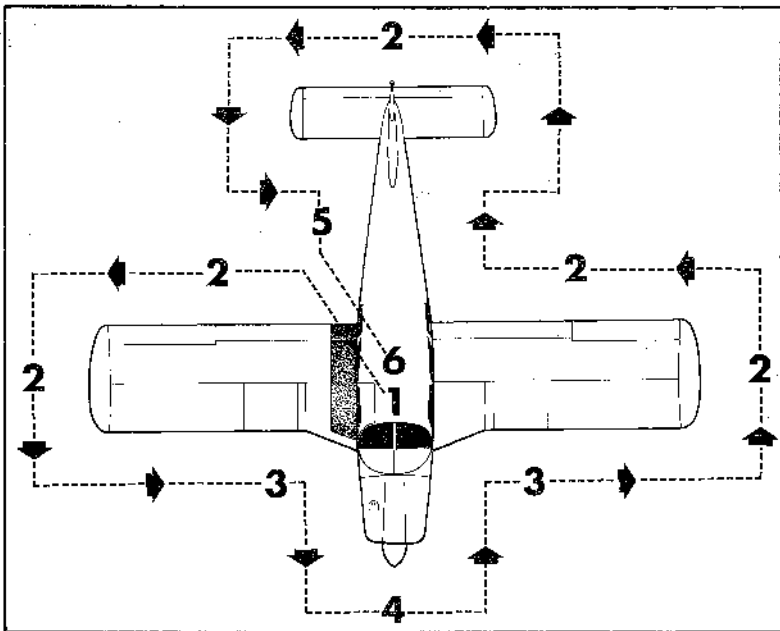
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SECTION III

OPERATING INSTRUCTIONS

PREFLIGHT

1. Master switch and ignition OFF.
2. a. Check for external damage and operational interference of control surfaces or hinges.  
 b. Insure that wings and control surfaces are free of snow, ice or frost.
3. a. Visually check fuel supply and secure caps.





- b. Drain fuel tank sumps (two).
- c. Drain fuel system sump (left side of aircraft).
- d. Check that fuel system vents are open.
- e. Check main landing gear shock struts for proper inflation (approximately 4.50 inches showing).
- f. Check tires for cuts, wear and proper inflation.
- 4. a. Check windshield for cleanliness.
- b. Check propeller and spinner for defects or nicks.
- c. Check for obvious fuel or oil leaks.
- d. Check oil level (insure dipstick is properly seated).
- e. Check cowling and inspection covers for security.
- f. Check nose wheel tire for inflation and wear.
- g. Check nose gear shock strut for proper inflation (approximately 3.25 inches showing).
- h. Check for foreign matter in air inlet.
- 5. a. Stow tow-bar and control locks if used.
- b. Check baggage for storage and security.
- c. Close and secure the baggage compartment door.
- 6. a. Upon entering airplane, check that all primary flight controls operate properly.
- b. Close and secure cabin door.
- c. Check that required papers are in order and in the airplane.
- d. Fasten seat belts and shoulder harness. Check function of inertia reel.

### STARTING ENGINE

1. Set brakes ON.
2. Set the carburetor heat control in the full COLD position.
3. Select the desired tank with fuel selector valve.

#### Starting Engine When Cold:

1. Open throttle approximately 1/4 inch.
2. Turn the master switch ON.
3. Turn the electric fuel pump ON.
4. Move the mixture control to FULL RICH.

5. Engage the starter by rotating magneto switch clockwise and pressing in.

6. When the engine fires, advance throttle to desired setting. If the engine does not fire within five to ten seconds, disengage starter and prime with one to three strokes of the priming pump. Repeat the starting procedure.

#### Starting Engine When Hot:

1. Open the throttle approximately 1/2 inch.
2. Turn the master switch ON.
3. Turn the electric fuel pump ON.
4. Put mixture control in IDLE CUT-OFF.
5. Engage the starter by rotating magneto switch clockwise and pressing in. When the engine fires, advance the mixture control and move the throttle to desired setting.

#### Starting Engine When Flooded:

1. Open the throttle full.
2. Turn the master switch ON.
3. Turn the electric fuel pump OFF.
4. Put mixture control in IDLE CUT-OFF.
5. Engage the starter by rotating magneto switch clockwise and pressing in. When the engine fires, advance the mixture control and retard the throttle.

#### Starting With External Power Source:

An optional feature known as Piper External Power (PEP) allows the operator to use an external battery to crank the engine without having to gain access to the aircraft battery.

The procedure is as follows:

1. Turn aircraft MASTER SWITCH to OFF.
2. Connect RED lead of PEP kit jumper cable to POSITIVE (+) terminal of external 12 volt battery and BLACK lead to NEGATIVE (-) terminal.
3. Insert plug of jumper cable into socket located on aircraft fuselage.
4. Turn aircraft MASTER SWITCH to ON and proceed with

NORMAL engine starting technique.

5. After engine has been started, turn MASTER SWITCH to OFF and remove jumper cable plug from aircraft.

6. Turn aircraft MASTER SWITCH to ON and check alternator ammeter for indication of output. DO NOT ATTEMPT FLIGHT IF THERE IS NO INDICATION OF ALTERNATOR OUTPUT.

When the engine is firing evenly, advance the throttle to 800 RPM. If oil pressure is not indicated within thirty seconds; stop the engine and determine the trouble. In cold weather it will take a few seconds longer to get an oil pressure indication. If the engine has failed to start, refer to the "Lycoming Operating Handbook, Engine Troubles and Their Remedies."

Starter manufacturers recommend that cranking periods be limited to thirty seconds with a two minute rest between cranking periods. Longer cranking periods will shorten the life of the starter.

## WARM-UP AND GROUND CHECK

Warm-up the engine at 800 to 1200 RPM for not more than two minutes in warm weather, four minutes in cold weather. Avoid prolonged idling at low RPM as this practice may result in fouled spark plugs. If necessary to hold before take-off, it is recommended that the engine be idled at 1200 RPM.

The magnetos should be checked at 1800 RPM and the drop off on either magneto should not exceed 125 RPM and should be within 50 RPM of the other. Prolonged operation on one magneto should be avoided.

Check vacuum gauge, indicator should read 5" Hg  $\pm$  .1" Hg at 2000 rpm.

Check both the oil temperature and pressure. The temperature may be low for some time if the engine is being run for the first time of the day, but as long as the pressure is within limits the engine is ready for take-off.

Carburetor heat should also be checked prior to take-off to be sure that the control is operating properly and to clear any ice which may have formed during taxiing. Avoid prolonged ground operation with carburetor heat ON as the air is unfiltered.

The electric fuel pump should be turned off momentarily during ground check to make sure that the engine-driven pump is operating. The electric fuel pump should be on during take-off to prevent loss of power should the engine-driven pump fail. The engine is warm enough for take-off when the throttle can be opened without the engine faltering. Do not take off with a dead battery as some voltage is needed to excite the alternator.

**TAKE-OFF**

Just before take-off the following items should be checked:

- 1. Fuel - on proper tank
- 2. Electric fuel pump - ON
- 3. Engine gauges - checked
- 4. Carburetor heat - OFF
- 5. Mixture - RICH
- 6. Quadrant friction knob - set
- 7. Flaps - set
- 8. Trim tab - set
- 9. Controls - free
- 10. Door - latched
- 11. Safety belts/shoulder harness - fastened

43-52 KT

The take-off technique is conventional for the Cherokee. The tab should be set slightly aft of neutral, with the exact setting determined by the loading of the aircraft. Allow the airplane to accelerate to 50 to 60 MPH, then ease back on the wheel enough to let the airplane fly itself off the ground. Premature raising of the nose, or raising it to an excessive angle will result in a delayed take-off. After take-off let the aircraft accelerate to the desired climb speed by lowering the nose slightly.

Take-offs are normally made with flaps up. However, for short field take-offs, and for take-offs under difficult conditions such as deep grass or on a soft surface, distances can be reduced appreciably by lowering flaps to 25°. Two notches

**CLIMB**

The best rate of climb at gross weight will be obtained at 85 MPH. The best angle of climb may be obtained at 74 MPH. At 74 KT 64 KT

lighter than gross weight these speeds are reduced somewhat. For climbing en route a speed of 100 MPH is recommended. This will produce better forward speed and increased visibility over the nose during the climb. 87 KT

## STALLS

All controls are effective at speeds down through the stalling speed, and stalls are gentle and easily controlled.

Stall speed chart on following page is at gross weight. Stall speeds at lower weights will be correspondingly less.

STALL SPEED TABLE		
Angle of Bank	Flaps 40°	Flaps Retracted
0°	57 MPH 50 KT	67 MPH 58 KT
20°	59 MPH 51 KT	69 MPH 60 KT
40°	65 MPH 56 KT	77 MPH 67 KT
60°	81 MPH 70 KT	95 MPH 83 KT
Power Off – Gross Weight 2400 lbs.		

## CRUISING

The cruising speed is determined by many factors including power setting, altitude, temperature, loading and equipment installed on the airplane.

The normal cruising power is 75% of the rated horsepower of the engine. True airspeeds which may be obtained at various altitudes and power settings can be determined from the charts in Section IV of this handbook.

Use of the mixture control in cruising flight reduces fuel

consumption significantly, especially at higher altitudes. The mixture should be leaned during cruising operation above 5000 feet altitude and at pilot's discretion at lower altitudes when 75% power or less is being used. If any doubt exists as to the amount of power being used, the mixture should be in the FULL RICH position for all operations under 5000 feet.

To lean the mixture, pull the mixture control until the engine becomes rough, indicating that the lean mixture limit has been reached in the leaner cylinders. Then enrich the mixture by pushing the control towards the instrument panel until engine operation becomes smooth. The fuel flow meter will give a close approximation of the fuel being consumed.

If the airplane is equipped with the optional exhaust gas temperature (EGT) gauge, a more accurate means of leaning is available to the pilot. For best power mixture, lean the mixture until the peak EGT is reached, then enrich the mixture until the temperature drops a minimum of 25°F. For best economy mixture, lean until the peak EGT is reached and continue to lean until the temperature drops a minimum of 25°F. Do not lean above 75% power.

In order to keep the airplane in best lateral trim during cruising flight, the fuel should be used alternately from each tank. It is recommended that one tank be used for one hour after take-off, then the other tank be used for two hours, then return to the first tank, which will have approximately one and one half hours of fuel remaining if the tanks were full at take-off. The second tank will contain approximately one half hour of fuel. Do not run tanks completely dry in flight.

## APPROACH AND LANDING

Before landing check list:

100 KT

- |                            |   |
|----------------------------|---|
| 1. Fuel - on proper tank   | 4. Flaps - set (115 MPH MAX)              |
| 2. Electric fuel pump - ON | 5. Seat belts/shoulder harness - fastened |
| 3. Mixture - set           |   |

75 KT  
100 KT  
Full Flap  
65 KT

The airplane should be trimmed to an approach speed of about 85 MPH with flaps up. The flaps can be lowered at speeds up to 115 MPH, if desired, and the approach speed reduced 3 MPH for each additional notch of flaps. Carburetor heat should not be applied unless there is an indication of carburetor icing, since the use of carburetor heat causes a reduction in power which may be critical in case of a go-around. Full throttle operation with heat on is likely to cause detonation.

The amount of flap used during landings and the speed of the aircraft at contact with the runway should be varied according to the landing surface and conditions of wind and airplane loading. It is generally good practice to contact the ground at minimum possible safe speed consistent with existing conditions.

Normally, the best technique for short and slow landings is to use full flap and enough power to maintain the desired airspeed and approach flight path. Reduce the airspeed during flare out and contact the ground close to stalling speed. After ground contact hold the nose wheel off as long as possible. As the airplane slows down, drop the nose and apply the brakes. There will be less chance of skidding the tires if the flaps are retracted before applying the brakes. Braking is most effective when back pressure is applied to the control wheel; putting most of the airplane weight on the main wheels. In high wind conditions, particularly in strong crosswinds, it may be desirable to approach the ground at higher than normal speeds with partial or no flaps.

### STOPPING ENGINE

At the pilot's discretion, the flaps should be raised and the electric fuel pump turned off. After parking, the radios should be turned off and the engine stopped by pulling the mixture control to idle cut-off. The throttle should be left full aft to avoid engine vibration while stopping. Then the magneto and master switches should be turned off and the parking brake set.

## ENGINE POWER LOSS

The most common cause of engine power loss is mismanagement of the fuel. Therefore, the first step to take after engine power loss is to move the fuel selector valve to the tank not being used. This will often keep the engine running even if there is no apparent reason for the engine to stop on the tank being used.

If changing to another tank does not restore the engine:

1. Check fuel pressure and turn on electric fuel pump if off.
2. Push mixture control to full "RICH."

3. Check ignition switch. Turn to best operating magneto - left, right, or both.

## MOORING

The Cherokee should be moved on the ground with the aid of the nose wheel tow-bar provided with each plane and secured in the baggage compartment. Tie down ropes may be secured to rings provided under each wing and to the tail skid. The aileron and stabilator controls should be secured by looping the safety belt through the control wheel and pulling it tight. The rudder is held in position by its connections to the nose wheel steering and normally does not have to be secured. The flaps are locked when in the full up position and should be left retracted.

## WEIGHT AND BALANCE

It is the responsibility of the owner and pilot to determine that the airplane remains within the allowable weight vs. center of gravity envelope while in flight. For weight and balance data see the Airplane Flight Manual and Weight and Balance form supplied with each airplane.



## OPERATING TIPS

The following Operating Tips are of particular value in the operation of the Cherokee.

1. Learn to trim for take-off so that only a very light back pressure on the wheel is required to lift the airplane off the ground.

52 KT

2. The best speed for take-off is about 60 MPH under normal conditions. Trying to pull the airplane off the ground at too low an airspeed decreases the controllability of the airplane in event of engine failure.

100 KT

3. Flaps may be lowered at airspeeds up to 115 MPH. To reduce flap operating loads, it is desirable to have the airplane at a slower speed before extending the flaps.

4. Before attempting to reset any circuit breaker, allow a two to five minute cooling off period. Do not reset circuit breakers in flight

5. Before starting the engine, check that all radio switches, light switches, and the pitot heat switch are in the off position so as not to create an overloaded condition when the starter is engaged.

6. The overvoltage relay is provided to protect the electronics equipment from a momentary overvoltage condition (approximately 16.5 volts and up), or a catastrophic regulator failure. In the event of a momentary condition, the relay will open and the ammeter will indicate "0" output from the alternator. The relay may be reset by switching the master switch to "OFF" for approximately one minute and then returning the master switch to "ON." If after recycling the master switch the condition persists, the flight may be continued on battery alone by monitoring the voltmeter and reducing battery load to a minimum.

7. The vacuum gauge is provided to monitor the pressure available to assure the correct operating speed of the vacuum driven gyroscopic flight instruments. It also monitors the condition of the common air filter by measuring the flow of air through the filter.

If the vacuum gauge registers lower than  $5" \pm .10"$  Hg at 2000 RPM, the following items should be checked before flight:

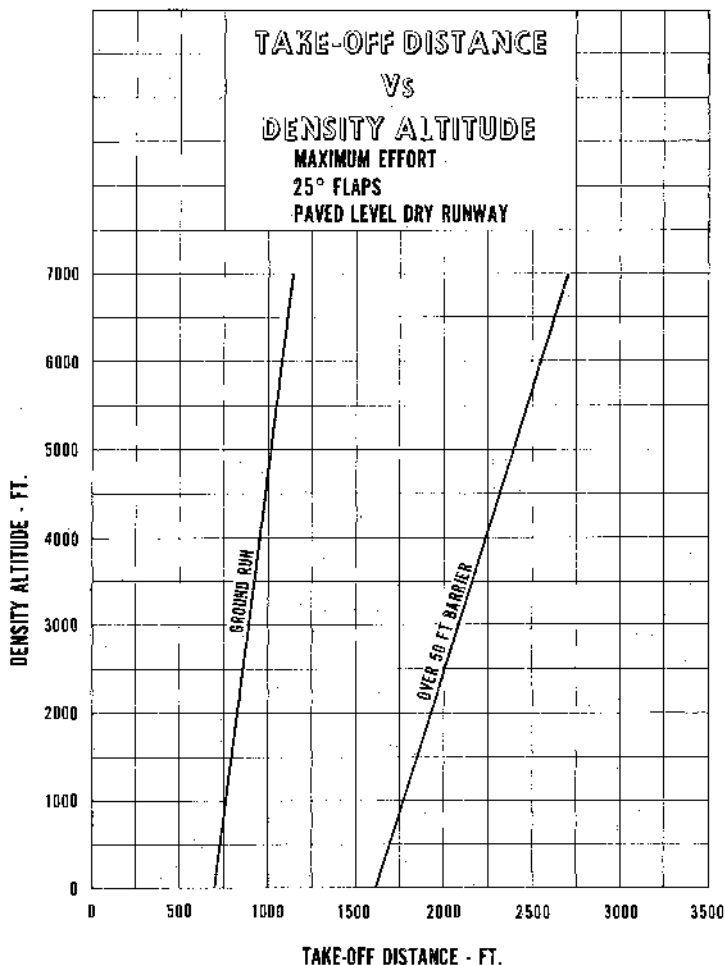
- a. Common air filter, could be dirty or restricted.

- b. Vacuum lines could be collapsed or broken.
- c. Vacuum pump, worn.
- d. Vacuum regulator, not adjusted correctly. The pressure, even though set correctly, can read lower under two conditions: (1) very high altitude, above 12,000 feet, (2) low engine rpm, usually on approach or during training maneuvers. This is normal and should not be considered a malfunction.

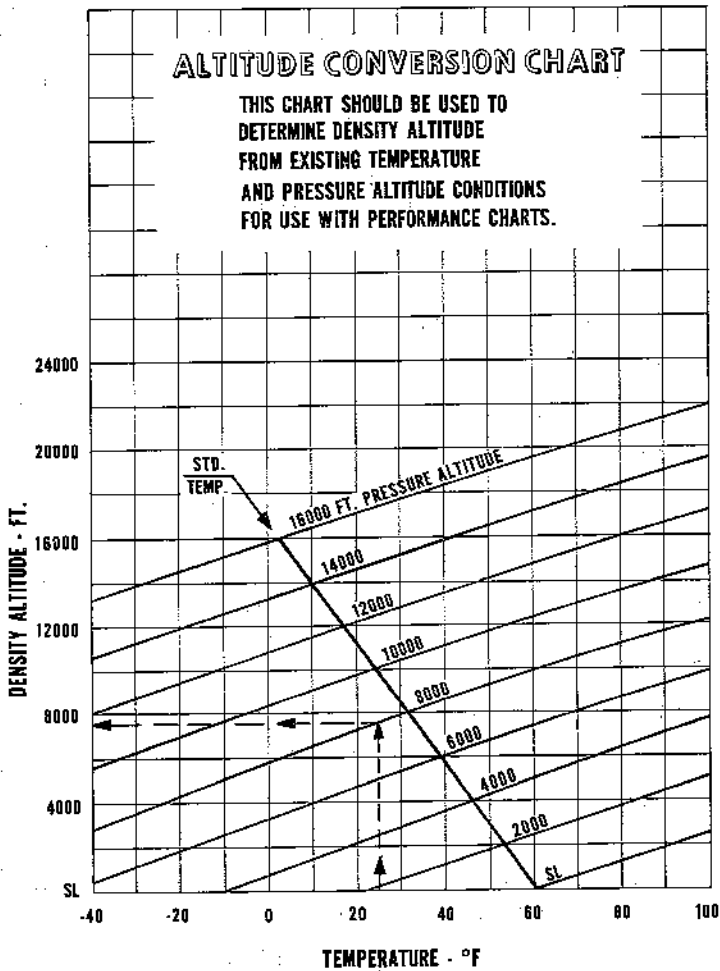
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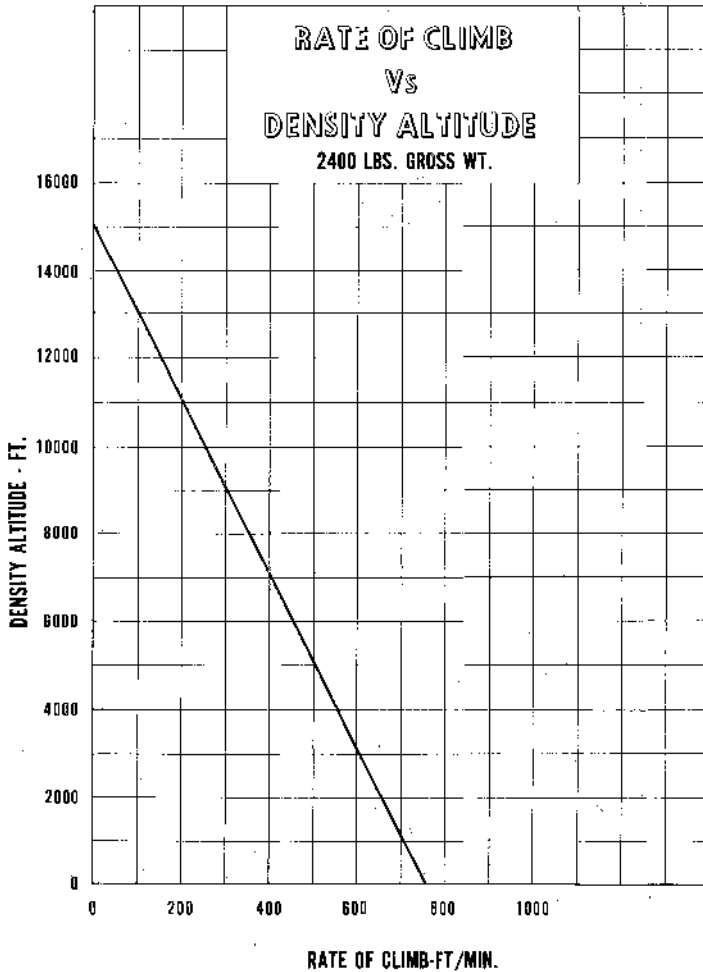
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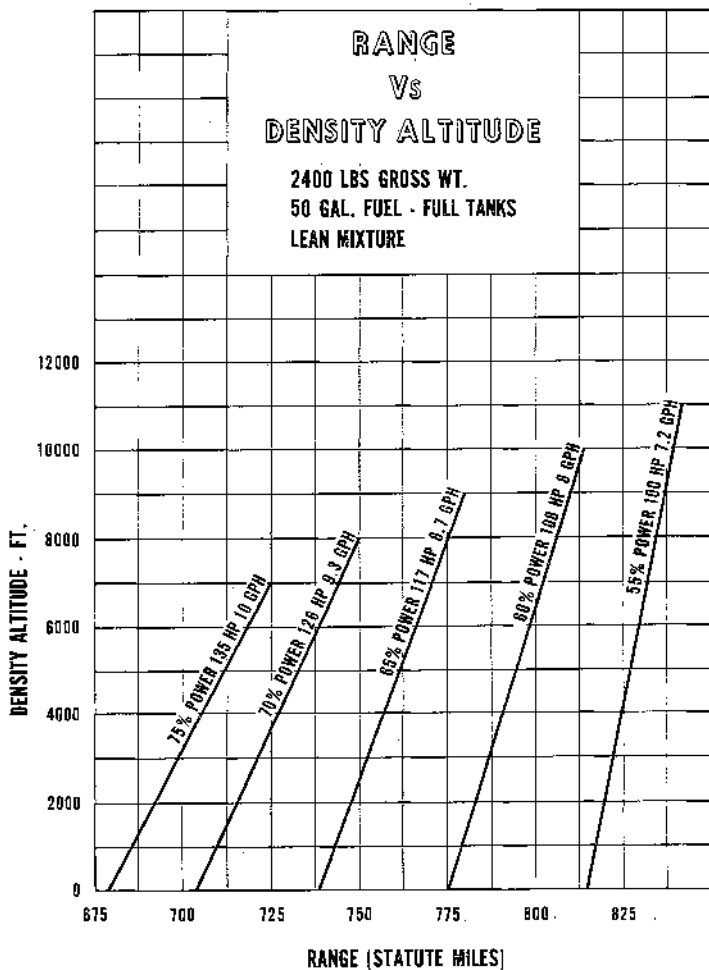
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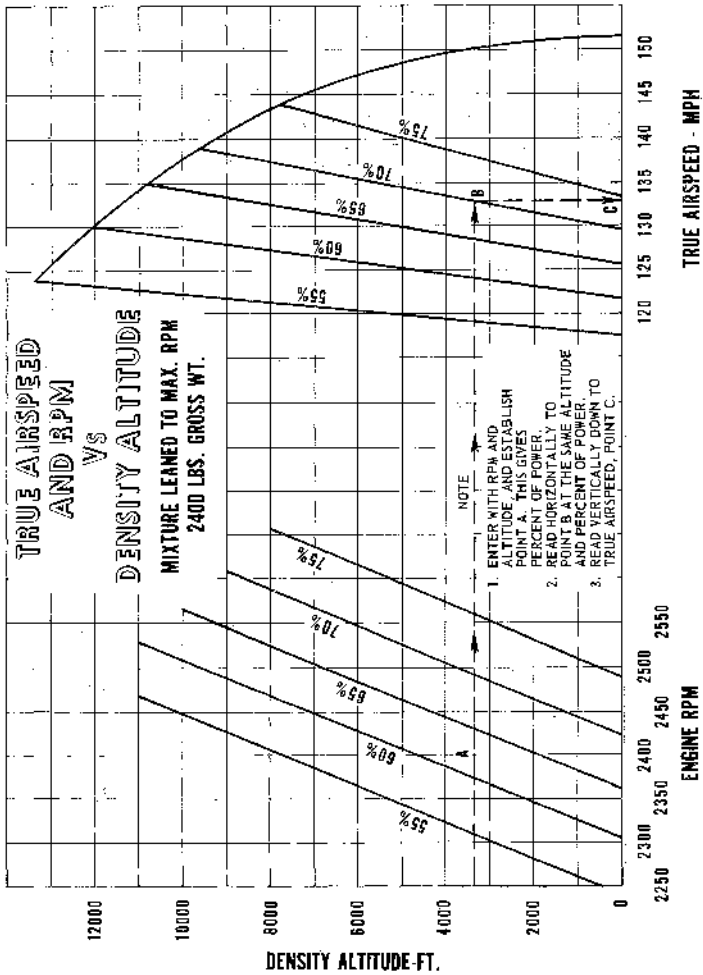
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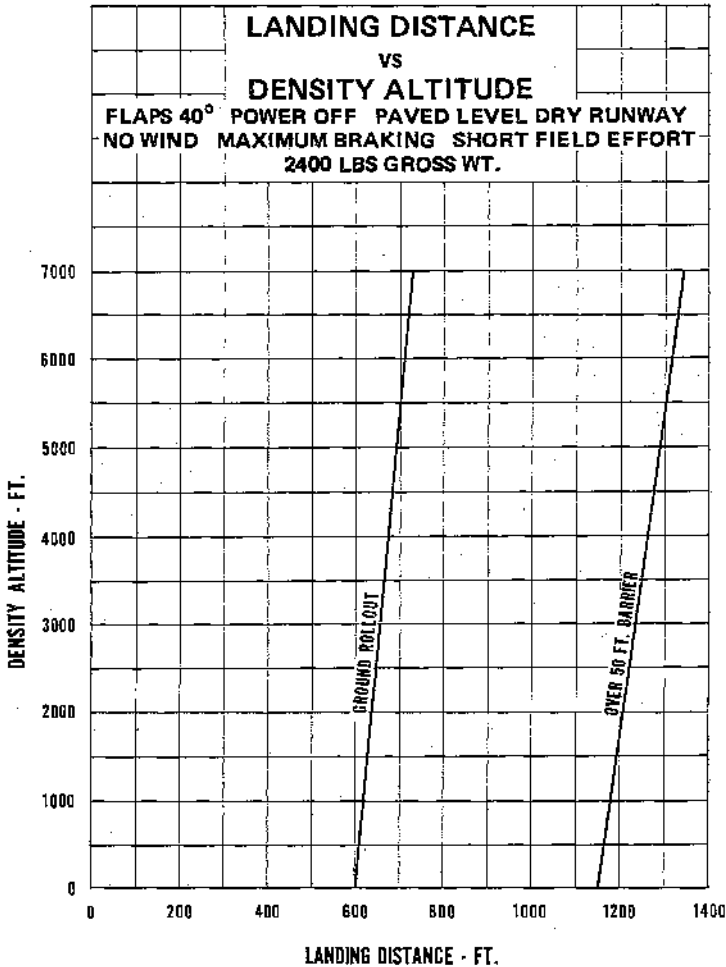


# PA-28-180 PIPER CHEROKEE





# PA-28-180 PIPER CHEROKEE



### Power Setting Table-Lycoming Model O-360 Series, 180 HP Engine

Press. Alt	Std Alt Temp °F	108 HP 60% Power RPM	117 HP 65% Power RPM	126 HP 70% Power RPM	135 HP 75% Power RPM	Press. Alt
SL	59	2290	2370	2440	2500	SL
1,000	55	2310	2390	2460	2520	1,000
2,000	52	2330	2410	2480	2540	2,000
3,000	48	2350	2430	2500	2560	3,000
4,000	45	2370	2450	2520	2580	4,000
5,000	41	2390	2470	2540	2600	5,000
6,000	38	2410	2490	2560	2620	6,000
7,000	34	2430	2510	2580	2640	7,000
8,000	31	2450	2530	2600	—	8,000
9,000	27	2470	2550	2620	—	9,000
10,000	23	2490	2570	—	—	10,000
11,000	19	2510	2590	—	—	11,000
12,000	16	2530	—	—	—	12,000

## SECTION V

### GENERAL MAINTENANCE

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## SECTION V

### GENERAL MAINTENANCE

This section of the Cherokee "D" Handbook contains information which pertains to minor maintenance of the airplane. For further maintenance assistance refer to the Service Manual for this airplane. Any complex repairs or modification should be accomplished by a Piper Certified Service Center or equivalent.

### LANDING GEAR SERVICE

The main wheels are Cleveland Aircraft Products, Model #40-86, with Cleveland single disk hydraulic brake assemblies, Model #30-55. The nose wheel is a Cleveland Aircraft Products, Model #38501. All wheels use a 6.00 x 6, four ply rating, type III tire with tube.

Main wheels are easily removed by taking off the hub cap, axle nut, and the two bolts holding the brake segment in place, after which the wheel slips easily from the axle.

Tires are demounted from the wheels by deflating the tire, removing the three through-bolts, and separating the wheel halves.

Landing gear oleo struts should be checked for proper strut exposures and fluid leaks. The required extensions for the strut when under normal static load (empty weight of airplane plus full fuel and oil) is 3.25 inches for the nose gear and 4.50 inches for the main gear. Should the strut exposure be below that required, it should be determined whether air or oil is required by first raising the airplane on jacks. Depress the valve core to allow

air to escape from the strut housing chamber. Remove the filler plug and slowly raise the strut to full compression. If the strut has sufficient fluid it will be visible up to the bottom of the filler plug hole and will then only require proper inflation.

Should fluid be below the bottom of the filler plug hole, oil should be added. Replace the plug with valve core removed, attach a clear plastic hose to the valve strut of the filler plug and submerge the other end in a container of hydraulic fluid (MIL-H-5606). Fully compress and extend the strut several times thus drawing fluid from the container and expelling air from the strut chamber. To allow the fluid to enter the bottom chamber of the main gear strut housing, the torque link assembly must be disconnected to let the strut be extended a minimum of 10 inches. (The nose gear torque links need not be disconnected.) Do not allow the strut to extend more than 12 inches. When air bubbles cease to flow through the hose, compress the strut fully and again check fluid level. Reinstall the valve core and filler plug, and the main gear torque links, if disconnected.

With fluid in the strut housing at the correct level, attach a strut pump to the air valve and with the airplane on the ground, inflate the oleo strut to the correct height.

In jacking the Cherokee for landing gear or other service, a jack kit (available through Piper Dealers or Distributors) should be used. This kit consists of two hydraulic jacks and a tail stand. At least 250 pounds of ballast should be placed on the base of the tail stand before the airplane is jacked up. The hydraulic jacks should be placed under the jack points on the bottom of the wing and the airplane jacked up until the tail skid is at the right height to attach the tail stand. After attaching the tail stand and adding the ballast, the jacking may be continued until the aircraft is at the height desired.

The steering arms from the rudder pedals to the nose wheel are adjusted at the rudder pedals or at the nose wheel by turning in or out the threaded rod end bearings. Adjustment is normally accomplished at the forward end of the rods and should be done in such a way that the nose wheel is in line with the fore and aft axis of the plane when the rudder pedals and rudder are centered.

Alignment of the nose wheel can be checked by pushing the airplane back and forth with the rudder centered to determine that the plane follows a perfectly straight line. The turning arc of the nose wheel is 22 degrees in either direction and factory adjusted at stops on the bottom of the forging. The turning radius of the nose wheel is 17 feet.

The steering arm stops should be carefully adjusted so that the nose wheel reaches its full travel just after the rudder hits its stops. This guarantees that the rudder will be allowed to move through its full travel.

### **BRAKE SERVICE**

The brake system is filled with MIL-H-5606 (Petroleum base) hydraulic brake fluid. This should be checked at every 50 hour inspection and replenished when necessary by filling the brake reservoir on the upper left front side of the firewall to the indicated level. If the system as a whole has to be refilled with fluid it should be done from the brake end of the system by filling with fluid under pressure. This will eliminate air from the system as it is being filled.

No adjustment of brake clearances is necessary on the Cherokee brakes. If after extended service the brake blocks become worn excessively, they are easily replaced with new segments.

### **TIRE INFLATION**

For maximum service from the tires on the Cherokee, keep the tires inflated to the proper pressure of 24 pounds for all three wheels. Interchange the tires on the main wheels if necessary to produce even wear. All wheels and tires are balanced before original installation, and the relationship of the tire, tube, and

wheel should be maintained if at all possible. Unbalanced wheels can cause extreme vibration on take-off. In the installation of new components it may be necessary to rebalance the wheel with the tire mounted.

### CARE OF WINDSHIELD AND WINDOWS

A certain amount of care is needed to keep the plexiglas windows clean and unmarred. The following procedure is recommended:

1. Flush with clean water and dislodge excess dirt, mud, etc. with your hand.
2. Wash with mild soap and water or Piper Plastic Cleaner. Use a soft cloth or sponge. Do not rub.
3. Remove oil, grease or sealing compounds with a soft cloth and kerosene.
4. After cleaning, apply a thin coat of hard polishing wax. Rub lightly with a soft cloth.
5. A severe scratch or mar may be removed by using jeweler's rouge to rub out the scratch, smoothing, and then applying wax.

### BATTERY SERVICE

Access for service or inspection of the battery is obtained through the removal of the panel at the right rear side of the baggage compartment. The stainless steel box has a plastic drain tube which is normally closed off with a clamp and which should be opened occasionally to drain off any accumulation of liquid. The battery should be checked for proper fluid level, but must not be filled above the baffle plates. Use only water - no acid. A hydrometer check should be performed to determine the percent of charge present in the battery.

If the battery is not up to charge, recharge starting at a 4 ampere rate and finishing with a 2 ampere rate. Quick charges are not recommended.

## FUEL AND OIL REQUIREMENTS

Aviation Grade 91/96 Octane (minimum) fuel must be used in the Cherokee. Because the use of lower grades can cause serious damage in a very short period of time, the engine warranty is invalidated by such use.

The oil capacity of the Lycoming O-360-A4A is 8 quarts, and the minimum safe quantity is 2 quarts. It is recommended that the oil and oil filter be changed every 50 hours, or sooner under unfavorable conditions. The following grades are recommended for the specific temperatures:

Temperatures above 60° F	S.A.E. 50
Temperatures between 30° and 90° F	S.A.E. 40
Temperatures between 0° and 70° F	S.A.E. 30
Temperatures below 10° F	S.A.E. 20

## FUEL SYSTEM

The fuel screen in the strainer will require cleaning every 50 hour inspection. The strainer, located ahead of the firewall, is accessible for cleaning by removal of the lower cowl. When the strainer is reassembled after cleaning, a small amount of grease applied to the gasket will facilitate assembly.



## CARE OF AIR FILTER

The carburetor air filter must be cleaned at least once every fifty hours. Under extremely adverse conditions of operation it may be necessary to clean the filter daily. Extra filters are inexpensive and a spare should be kept on hand and used as a rapid replacement.

The filter manufacturer recommends that the filter be tapped gently to remove dirt particles. Do not blow out with compressed air.

## LEVELING AND RIGGING

Leveling the Cherokee "D" for purposes of weighing or rigging is accomplished as follows:

1. Partially withdraw two machine screws located immediately below the left front side window. These screws are leveling points, and the airplane is longitudinally level when a level placed on the heads of these screws indicates level.

2. To put the airplane in a longitudinally level position on scales, first block the main gear oleos in the fully extended position, then deflate the nose wheel tire until the proper attitude is obtained. For rigging only, the airplane may be placed on jacks for leveling.

3. To level the airplane laterally, place a level across the baggage compartment floor along the rear bulkhead.

Rigging: Although the fixed flight surfaces on the Cherokee cannot be adjusted for rigging purposes, it may be necessary upon occasion to check the position of these surfaces. The movable surfaces all have adjustable stops, as well as adjustable turnbuckles on the cables or push-pull tubes, so that their range of travel can be altered. The positions and angular travels of the various surfaces are as follows:

1. Wings: 7° dihedral, 2° washout.
2. Stabilator Travel: 18° up, 2° down, tolerance  $\pm 1^\circ$ .

3. Fin should be vertical and in line with center of fuselage.
4. Aileron Travel: 30° up, 15° down, tolerance  $\pm 2^\circ$ .
5. Flap Travel: 10°, 25°, 40°, tolerance  $\pm 2^\circ$ .
6. Rudder Travel: 27° right and left, tolerance  $\pm 2^\circ$ .
7. Stabilator Tab Travel: 3° up, 12° down, tolerance  $\pm 1^\circ$ .

Cable tensions for the various controls are as follows:

Rudder: 40 $\pm$ 5 lbs.	Stabilator: 40 $\pm$ 5 lbs.
Ailerons: 40 $\pm$ 5 lbs.	Stabilator Trim: 10 $\pm$ 1 lbs.
Flaps: Approx. 10 lbs.	

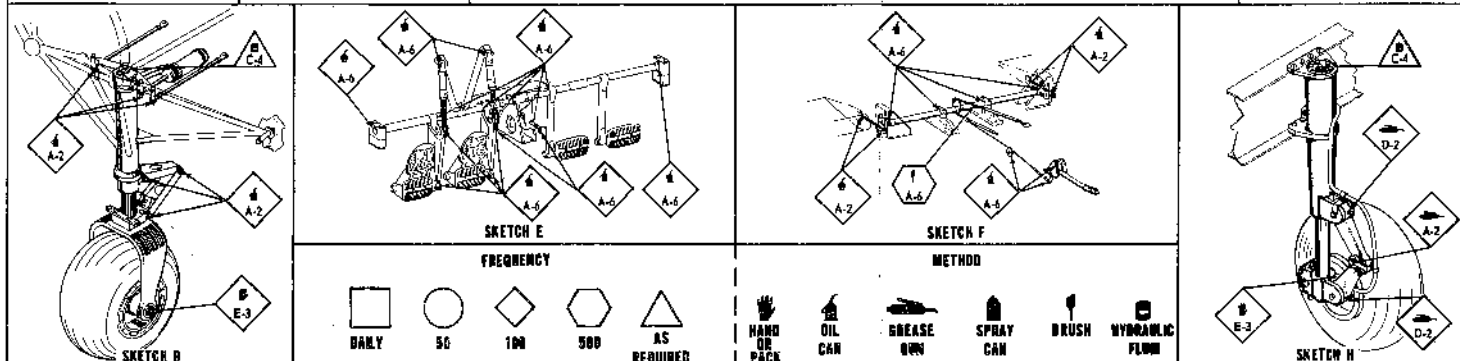
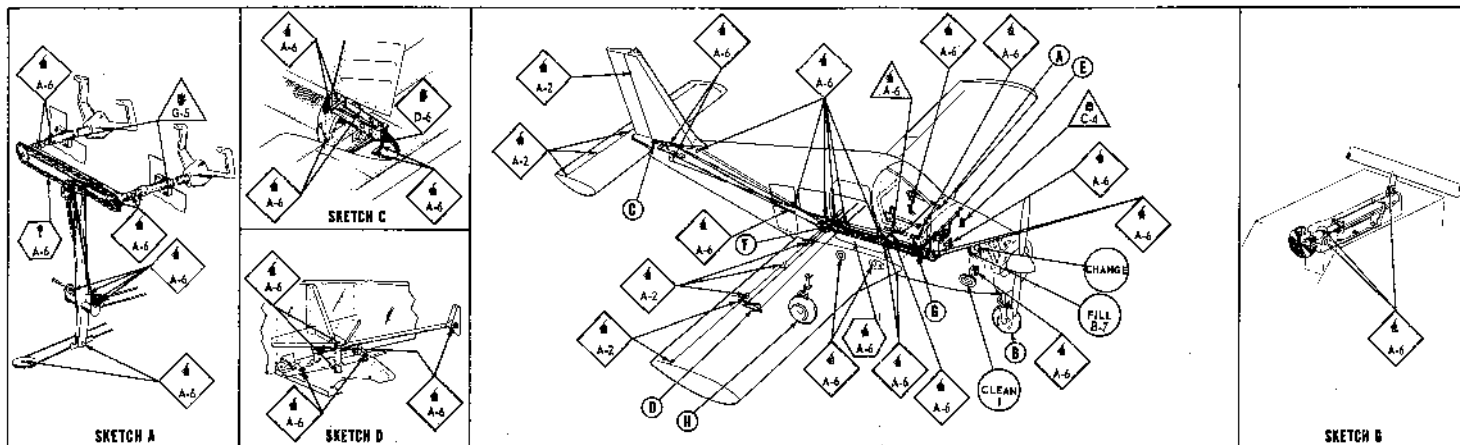
For extreme cases of wing heaviness, the flap on the wing heavy side may be adjusted down from the zero position as desired.

The service manual should be consulted for the proper method of adjusting surface travels.

## SERIAL NUMBER PLATE

The serial number plate is located near the stabilator on the left side of the airplane. Refer to this number for service or warranty matters.

NOTES



EXAMPLE	TYPE OF LUBRICANTS		SPECIAL INSTRUCTIONS	NOTES	
<p><b>LUBRICATION CHART PA-28-180 "D"</b></p>	IDENTIFICATION LETTER	SPECIFICATION	LUBRICANT		
	A	MIL-L-7870	LUBRICATING OIL, GENERAL PURPOSE, LOW TEMPERATURE	1. AIR FILTER - TO CLEAN FILTER, TAP GENTLY TO REMOVE DIRT PARTICLES. DO NOT BLOW OUT WITH COMPRESSED AIR OR USE OIL. REPLACE FILTER IF PUNCTURED OR DAMAGED.	1. PILOT AND PASSENGER SEATS - LUBRICATE TRACK ROLLERS AND STOP PINS AS REQUIRED. (TYPE OF LUBRICANT: "A")
	B	MIL-L-6082	LUBRICATING OIL, AIRCRAFT RECIPROCATING ENGINE (PISTON) GRADE AS SPECIFIED SAE 50 ABOVE 60° F AIR TEMP. SAE 40 30° TO 50° F AIR TEMP. SAE 30 0° TO 20° F AIR TEMP. SAE 20 BELOW 10° F AIR TEMP.	2. BEARINGS AND BUSHINGS - CLEAN EXTERIOR WITH A DRY TYPE SOLVENT BEFORE LUBRICATING.	2. WHEEL BEARINGS REQUIRE CLEANING AND REPACKING AFTER EXPOSURE TO AN ABNORMAL QUANTITY OF WATER.
	C	MIL-H-5606	HYDRAULIC FLUID, PETROLEUM BASE	3. WHEEL BEARINGS - DISASSEMBLE AND CLEAN WITH A DRY TYPE SOLVENT. ASCERTAIN THAT GREASE IS PACKED BETWEEN THE BEARING ROLLER AND CONE. DO NOT PACK GREASE IN WHEEL HOUSING.	3. FUEL SELECTOR VALVE - LUBRICATE FUEL SELECTOR VALVE AS REQUIRED. REFER TO PIPER SERVICE LETTER NO. 33.
	D	MIL-G-23827	GREASE, AIRCRAFT AND INSTRUMENT, GEAR AND ACTUATOR SCREW	4. OILED STRUTS AND BRAKE RESERVOIR - FILL PER INSTRUCTIONS ON UNIT OR CONTAINER, OR REFER TO SERVICE MANUAL, SECTION II.	4. SEE LYCOWING SERVICE INSTRUCTIONS NO. 1014 FOR USE OF DETERGENT OIL.
	E	MIL-G-3545	GREASE, AIRCRAFT, HIGH TEMPERATURE	5. "D" RING, CONTROL SHAFT BUSHING - DISASSEMBLY "O" RING RETAINER PLATES FROM INSTRUMENT PANEL, LUBRICATE "O" RING AND REASSEMBLE.	
	F	DUPONT ALL PURPOSE SLIP SPRAY #6611 OR EQUIVALENT		6. LUBRICATION POINTS - WIPE ALL LUBRICATION POINTS CLEAN OF OLD GREASE, OIL, DIRT, ETC. BEFORE LUBRICATING.	
G	PARKER "D" RING LUBRICANT		7. INTERVALS BETWEEN OIL CHANGES CAN BE INCREASED AS MUCH AS 100% ON ENGINES EQUIPPED WITH FULL FLOW (CARTRIDGE TYPE) OIL FILTERS - PROVIDED THE ELEMENT IS REPLACED EACH 50 HOURS OF OPERATION.		

This section was taken from a different PA-28-180 manual

**SECTION IV**

**EMERGENCY PROCEDURES**

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**SECTION IV****EMERGENCY PROCEDURES****INTRODUCTION**

This section contains procedures that are recommended if an emergency condition should occur during ground operation, takeoff, or in flight. These procedures are suggested as the best course of action for coping with the particular condition described, but are not a substitute for sound judgement and common sense. Since emergencies rarely happen in modern aircraft, their occurrence is usually unexpected, and the best corrective action may not always be obvious. Pilots should familiarize themselves with the procedures given in this section and be prepared to take appropriate action should an emergency arise.

Most basic emergency procedures, such as power off landings, are a normal part of pilot training. Although these emergencies are discussed herein, this information is not intended to replace such training, but only to provide a source of reference and review, and to provide information on procedures which are not the same for all aircraft. It is suggested that the pilots review standard emergency procedures periodically to remain proficient in them.

**ENGINE POWER LOSS DURING TAKEOFF**

The proper action to be taken if loss of power occurs during takeoff will depend on circumstances.

1. If sufficient runway remains for a normal landing, land straight ahead.
2. If insufficient runway remains, maintain a safe airspeed and make only a shallow turn if necessary to avoid obstructions. Use of flaps depends on circumstances. Normally, flaps should be fully extended for touchdown.

3. If you have gained sufficient altitude to attempt a restart, proceed as follows:

- a. MAINTAIN SAFE AIRSPEED
- b. FUEL SELECTOR - SWITCH TO ANOTHER TANK CONTAINING FUEL
- c. ELECTRIC FUEL PUMP - CHECK ON
- d. MIXTURE - CHECK RICH
- e. CARBURETOR HEAT - ON

#### **NOTE**

If engine failure was caused by fuel exhaustion, power will not be regained after tanks are switched until empty fuel lines are filled, which may require up to ten seconds.

If power is not regained, proceed with the POWER OFF LANDING procedure.

### **ENGINE POWER LOSS IN FLIGHT**

Complete engine power loss is usually caused by fuel flow interruption, and power will be restored shortly after fuel flow is restored. If power loss occurs at low altitude, the first step is to prepare for an emergency landing (See POWER OFF LANDING). Maintain an airspeed of at least 82 MPH IAS, and if altitude permits, proceed as follows:

**71 KT**

1. Fuel Selector - Switch to another tank containing fuel
2. Electric Fuel Pump - On
3. Mixture - Rich
4. Carburetor Heat - On
5. Engine Gauges - Check for an indication of the cause of power loss
6. Primer - Check Locked
7. If no fuel pressure is indicated, check tank selector position to be sure it is on a tank containing fuel.

When power is restored:

8. Carburetor Heat - Off
9. Electric Fuel Pump - Off

If the above steps do not restore power, prepare for an emergency landing.

If time permits:

1. Ignition Switch - "L" then "R" then back to "BOTH."
2. Throttle and Mixture - Different settings. (This may restore power if problem is too rich or too lean a mixture, or a partial fuel system restriction.)
3. Try another fuel tank. (Water in the fuel could take some time to be used up, and allowing the engine to windmill may restore power. If power loss is due to water, fuel pressure indications will be normal.)

#### **NOTE**

If engine failure was caused by fuel exhaustion, power will not be regained after tanks are switched until empty fuel lines are filled, which may require up to ten seconds.

If power is not restored, proceed with **POWER OFF LANDING** procedures.

#### **POWER OFF LANDING**

71 KT If loss of power occurs at altitude, trim the aircraft for best gliding angle, 82 MPH IAS, and look for a suitable field. If measures taken to restore power are not effective, and if time permits, check your charts for airports in the immediate vicinity; it may be possible to land at one if you have sufficient altitude. If possible, notify the FAA by radio of your difficulty and intentions. If another pilot or passenger is aboard, let them help.



When you have located a suitable field, establish a spiral pattern around this field. Try to be at 1000 feet above the field at the downwind position, to make a normal approach. Excess altitude may be lost by widening your pattern, using flaps or slipping, or a combination of these.

Touchdown should normally be made at the lowest possible airspeed, with full flaps.

When committed to landing:

1. Ignition - Off
2. Master Switch - Off
3. Fuel Selector - Off
4. Mixture - Idle Cut-Off
5. Seat belt - Tight

## **FIRE**

The presence of fire is noted through smoke, smell, and heat in the cabin. It is essential that the source of the fire be promptly identified through instrument readings, character of the smoke, or other indications, since the action to be taken differs somewhat in each case.

Source of fire - Check

1. Electrical Fire (smoke in cabin):
  - a. Master Switch - Off
  - b. Vents - Open
  - c. Cabin Heat - Off
  - d. Land as soon as possible.

2. Engine Fire In Flight:
  - a. Fuel Selector - Off
  - b. Throttle - Closed
  - c. Mixture - Idle cut-off
  - d. Heater - Off (In all cases of fire)
  - e. Defroster - Off (In all cases of fire)
  - f. If terrain permits, land immediately.

The possibility of an engine fire in flight is extremely remote. The procedure given above is general and pilot judgment should be the deciding factor for action in such an emergency.

3. Engine Fire During Start:
  - a. If engine has not started
    - (1) Mixture - Idle cut-off
    - (2) Throttle - Open
    - (3) Turn engine with starter (This is an attempt to pull the fire into the engine.)
  - b. If engine has already started and is running, continue operating to try pulling the fire into the engine.
  - c. In either case stated a. and b., if the fire continues longer than a few seconds, the fire should be extinguished by the best available external means.
  - d. If external fire extinguishing is to be applied:
    - (1) Fuel Selector - Off
    - (2) Mixture - Idle cut-off

Engine fires during start are usually the result of over priming. The procedure above is designed to draw the excess fuel back into the induction system.

## **LOSS OF OIL PRESSURE**

Loss of oil pressure may be either partial or complete. A partial loss of oil pressure usually indicates a malfunction in the oil pressure regulating system, and a landing should be made as soon as possible to investigate the cause and prevent engine damage.

A complete loss of oil pressure indication may signify oil exhaustion or may be the result of a faulty gauge. In either case, proceed toward the nearest airport, and be prepared for a forced landing. If the problem is not a pressure gauge malfunction, the engine may stop suddenly. Maintain altitude until such time as a dead stick landing can be accomplished. Don't change power settings unnecessarily, as this may hasten complete power loss.

Depending on the circumstances, it may be advisable to make an off airport landing while power is still available, particularly if other indications of actual oil pressure loss, such as sudden increases in temperatures, or oil smoke, are apparent, and an airport is not close.

If engine stoppage occurs, proceed to **POWER OFF LANDING**.

### **LOSS OF FUEL PRESSURE**

1. Electric Boost Pump - On
2. Fuel Selector - Check on Full Tank

If problem is not an empty fuel tank, land as soon as practical, and have engine-driven fuel pump checked.

### **HIGH OIL TEMPERATURE**

An abnormally high oil temperature indication may be caused by a low oil level, an obstruction of the oil cooler, damaged or improper baffle seals, a defective gauge, or other causes. Land as soon as practical at an appropriate airport and have the cause investigated.

A steady, rapid rise in oil temperature is a sign of trouble. Land at the nearest airport and let a mechanic investigate the problem. Watch the oil pressure gauge for an accompanying loss of pressure.

**ALTERNATOR FAILURE**

Loss of alternator output is detected through a zero reading on the ammeter. Before executing the following procedure, insure that the reading is zero and not merely low by actuating an electrically powered device, such as the landing light. If no increase in the ammeter reading is noted, alternator failure can be assumed.

1. Reduce Electrical Load.
2. Alternator Circuit Breakers - Check
3. "Alt" Switch - Off (for 1 second), Then On

If the ammeter continues to indicate no output, or alternator will not stay reset, turn off "Alt" switch, maintain minimum electrical load and land as soon as practical. All electrical load is being supplied by the battery.

**ENGINE ROUGHNESS**

Engine roughness is usually due to carburetor icing, which is indicated by a drop in RPM, and may be accompanied by a slight loss of airspeed or altitude. If too much ice is allowed to accumulate, restoration of full power may not be possible; therefore, prompt action is required.

Carburetor Heat - On (See Note) RPM will decrease slightly and roughness will increase. Wait for a decrease in engine roughness or an increase in RPM, indicating ice removal. If no change in approximately one minute, return carburetor heat to COLD. If the engine is still rough, try steps below.

1. Mixture - Adjust for maximum smoothness. Engine will run rough if too rich or too lean.
2. Electric Fuel Pump - On
3. Fuel Selector - Change tanks to see if fuel contamination is the problem.
4. Engine Gauges - Check for abnormal readings. If any gauge readings are abnormal, proceed accordingly.

5. Magneto Switch - "L" then "R," then back to "BOTH." If operation is satisfactory, on either magneto, proceed on that magneto at reduced power, with mixture full rich, to a landing at the first available airport.

If roughness persists, prepare for a precautionary landing at pilot's discretion.

### **NOTE**

Partial carburetor heat may be worse than no heat at all, since it may partially melt the ice, which will refreeze in the intake system. When using carburetor heat, therefore, always use full heat and when ice is removed, return the control to the full cold position.

### **SPINS**

Intentional spins are prohibited in this aircraft. If a spin is inadvertently entered, immediately use the following recovery procedures:

1. THROTTLE - IDLE
2. RUDDER - FULL OPPOSITE TO DIRECTION OF ROTATION
3. CONTROL WHEEL - FULL FORWARD
4. RUDDER - NEUTRAL (WHEN ROTATION STOPS)
5. CONTROL WHEEL - AS REQUIRED TO SMOOTHLY REGAIN LEVEL FLIGHT ATTITUDE

**OPEN DOOR**

The cabin door on the Cherokee is double latched, so the chance of its springing open in flight at both the top and bottom is remote. However, should you forget the upper latch, or not fully engage the lower latch, the door may spring partially open. This will usually happen at takeoff or soon afterward. A partially open door will not affect normal flight characteristics, and a normal landing can be made with the door open. If both upper and lower latches open, the door will trail slightly open, and airspeed will be reduced slightly.

To close a door in flight, proceed as follows:

1. Slow aircraft to 100 MPH IAS 87 KT
2. Cabin Vents - Close
3. Storm Window - Open
4. If upper latch is open - latch. If lower latch is open - open top latch, push door further open, and then close rapidly. Latch top latch. A slip in the direction of the open door will assist in latching procedure.